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SCOUT ERROR ANALYSIS PHASE II FINAL REPORT

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Prepared under Contract No. NAS1-6969 for NATIONAL AERONAUTICS AND SPACE ADMINISTRATION



SCOUT ERROR ANALYSIS PHASE II FINAL REPORT

By Lester B. Cohen 19 September 1968

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> Prepared under Contract No. NAS1-6969 TRW Systems Group Redondo Beach, California

> > for

NATIONAL AERONAUTICS AND SPACE ADMINISTRATION

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SUMMARY

This report is presented in fulfillment of the statement of work of contract No. NAS1-6969 for Phase II as amended on May 13, 1968. The statement of work required that TRW analyze the effects of Scout error sources, both individually and collectively, on the final trajectory conditions of the following two missions:

- (1) Reentry mission.
- (2) Five stage escape mission with both fourth and fifth stages spin stabilized.

TRW was to use the same methods and error sources developed under Phase I of the subject contract, with the Government supplying fifth stage data and error sources for the escape mission and new pitch profiles for both flights.

The results of analyzing the reentry mission produced the following one sigma errors at the time of nominal 400,000 feet altitude:

	nominal conditions	mean dispersion	standard <u>deviation</u>
Velocity	26202 ft/sec	-15.29 ft/sec	87.0 ft/sec
Flight Path Angle	-3.06 deg	.0736 deg	.553 deg
Inclination	50.425 deg	.047 deg	.432 deg

The analysis of the five stage escape mission resulted in the following dispersions at fifth stage burnout:

	$\frac{\texttt{nominal}}{\texttt{conditions}}$	mean dispersion	standard <u>deviation</u>
Velocity	39199.7 ft/sec	2.02 ft/sec	78.7 ft/sec
Altitude	107.56 nmi	.424 nmi	5.03 nmi
Flight Path Angle	11.769 deg	007 deg	•377 deg
Inclination	37.667 deg	.0168 deg	.0455 deg

INTRODUCTION

This report is presented in fulfillment of Phase II of contract No. NAS1-6969 as amended on May 13, 1968. The statement of work called for the analysis of two missions:

(1) A four stage trajectory designed to reenter the Earth's atmosphere.

TRW was instructed to determine the expected dispersions in the trajectory at an altitude of 400,000 feet.

(2) A five stage mission with the fifth stage spin stabilized designed to enter interplanetary space. The data for the fifth stage was supplied by the Scout Program Office at Langley Research Center.

The error source magnitudes and the method of analysis were to be the same as developed and utilized in Phase I of the subject contract. A brief description of these methods follows but the interested reader is referred to the Phase I final report (Reference 1) for a more detailed description of the scout vehicle.

The Scout simulation equations were assumed to be a quadratic form using coefficients previously calculated in the TRW N-Stage Trajectory Program (MVNS). There were three component parts:

Linear Errors:

$$\underline{\underline{X}}_{L} = \sum_{i=1}^{95} \underline{c}_{i} \Delta_{i}$$
linear
(1)

where $\underline{X}_{T_{i}}$ is the state vector at a stage time

 $\underline{\underline{c}}_{i}$ is the vector of linear sensitivity coefficients associated with the i^{th} error source

 $\boldsymbol{\Delta}_{\mbox{\scriptsize i}}$ is a random number with standard deviation equal to $\boldsymbol{\sigma}_{\mbox{\scriptsize i}}$

Non-linear Errors:

$$\underline{\underline{x}}_{NL} = \sum_{j=1}^{8} \underline{f}_{j}(\Delta_{j})$$
 (2)

where \underline{X}_{NT} is the non-linear state vector

 $\underline{f}_{j}(\Delta_{j})$ is the piece-wise linear function for the non-linear source j, formed by its effect at ± 10 , ± 20 , ± 30 .

Cross-term Errors:

$$\underline{\underline{X}}_{CT} = \sum_{k=1}^{11} \sum_{\ell=k+1}^{12} \underline{\underline{C}}_{k\ell} \Delta_k \Delta_{\ell}$$
(3)

where $\underline{\mathbf{X}}_{\mathrm{CT}}$ is the cross-term vector

 $\underline{c}_{\mathrm{k}\ell}$ is the cross-term sensitivity

finally:

$$\underline{X} = \underline{X}_{L} + \underline{X}_{NL} + \underline{X}_{CT} \tag{4}$$

The procedure is divided into three consecutive steps: (1) the generation of sensitivity coefficients of the state vector at the stage ignition times and at burnout or reentry with respect to error sources deviations; (2) the generation of cumulative distribution functions of desired orbital and reentry parameters using the sensitivity coefficients and the statistic of the error sources; (3) the analysis of these distribution functions. In Figure 1, a block diagram describes the relationship between the TRW programs which was used in the analysis.

Each step will be discussed separately in some detail.

Sensitivity Coefficient Generation

The state of any system, as a function of time, may be considered as a function of initial state as well as a number of performance parameters (thrust, commanded turning rates, aerodynamic properties, etc.). The effects caused by nominal values of these parameters serve to define a nominal state at some time of interest, t_k , i.e.,

$$X_N (t_k) = f(X_0, P, t_0; t_k)$$
 (5)

However, if these parameters are allowed to depart from their nominal values, a perturbed state vector is generated:

$$X (t_k) = f (X_o, P + \delta P, t_o; t_k)$$
 (6)

If the difference in the two state vectors is expanded into a Taylor series about the nominal value:

$$\delta X_{i} = \sum_{j=1}^{q} C_{ij} \delta P_{j} + \frac{1}{2} \sum_{\substack{k=1 \ j=1}}^{q} C_{ijk}^{(2)} \delta P_{j} \delta P_{k} + \dots \qquad i = 1, 2, \dots 6 \quad (7)$$

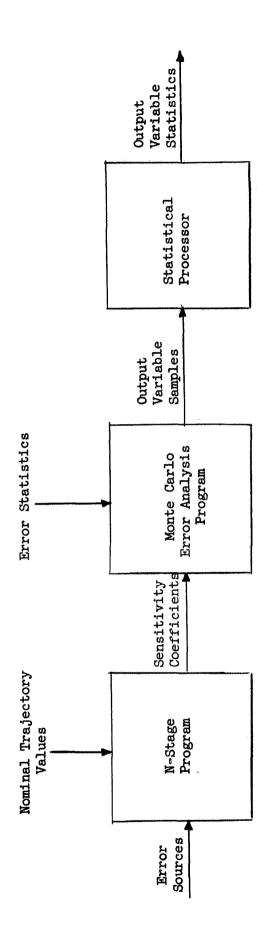


Figure 1. Programs Used in the Scout Error Analysis Program.

where:

 δX_i = the ith component of the difference between perturbed and nominal state vectors

q is the dimension of the performance variation vector

 C_{ij} is the partial derivative $\partial X_i/\partial P_j$

 $C_{ijk}^{(2)}$ is the second partial derivative $\partial^2 X_i/\partial P_j P_k$

The number of terms in the expansion is sufficient to adequately represent the functional dependence of the state variation with respect to these parameters. If this expansion is truncated to include only second order terms, there are q first order parameters and $\frac{q(q+1)}{2}$ second order coefficients. This would necessitate $\frac{q(q+3)}{2}$ computer runs which for the values of q modelled for the Scout analysis (~100) is unrealistic.

However, it is certainly true that certain error sources will contribute more to the state vector dispersion than others and only these major contributors need be investigated for nonlinear effects. Thus, if the linear effect of an error source on the dependent variables was found to be negligible in relation to that of more significant error sources, no attempt was made to investigate such minor sources in any greater detail. This procedure has been found by TRW Systems to be most efficient, since the time spent in modeling any error sources is roughly proportional to its overall contribution to the output. The procedure for generation of sensitivity coefficients was as follows:

1. Using the MVNS simulation of the nominal mission, perturb the system with a +3 sigma variation for each error source. Take the square root of the sum of the squares of the deviations and compare each deviation with the RSS total. If the individual deviation is less than 0.3 of the RSS total, the effect of the error source will be represented by a linear sensitivity function. This criterion guarantees that the error source deviation, when squared has a contribution of magnitude less than the total sum of the squares.

- 2. The selected sources were perturbed by ±10 and ±20 and ±30 to determine their degree of nonlinearity. Both postive and negative perturbations are required, since it is possible for the curve to be symmetrical while still nonlinear. If it were found that the sensitivity curve had a nonlinear shape, a straightline approximation was used to represent the partial as shown in Figure 2.
- 3. The next relationship to be found was the cross-correlations between these most significant variables. The difficulty of this problem is increased by the fact that the important combinations are not all known a priori. TRW Systems isolated and identified several combinations for the Phase I reference mission, so that it was not necessary to use the corresponding large amounts of machine time for each subsequent mission.

The non-linear sources investigated in this analysis were thrust misalignments and coning rate errors, since not only were these sources large contributors to the combined errors, but it is clear that major effects such as velocity loss will be the same whether the misalignment is positive or negative. Conversely, a large effect such as specific impulse tends to be linear, more impulse produces more velocity and vice-versa. In addition, this type of error source has no out of plane or in plane directional effects. To summarize, the non-linear sources used in the analysis of the reentry mission were:

Thrust misalignments, first stage (TMP1)
Thrust misalignments, first stage (TMY1)
Thrust misalignments, second stage (TMP2)
Thrust misalignments, second stage (TMY2)
Thrust misalignments, third stage (TMP3)
Thrust misalignments, third stage (TMY3)
Coning impulse, fourth stage (WC4P)
Coning impulse, fourth stage (WC4Y)

In the escape mission the following two sources were added to the list: coming impulse, fifth stage (WC5P), coming impulse, fifth stage (WC5Y)

The cross correlations used in the Scout analysis were those combining thrust misalignments at second and third stages with their respective control system errors. These combinations are necessary to discover any of the effects

of control system errors since if they are operating on a trajectory with no upsetting moments as second and third stage are, there will be no reason for any control system activity and the nominal burnout values will be reproduced. Therefore it was necessary to provide a large and relatively constant upsetting force which led logically to the choice of thrust misalignments. The pairings which were used in both missions were:

Second Stage:

Thrust Misalignment, Pitch: Rate Gain, Pitch Thrust Misalignment, Pitch: Dead Band, Pitch

Thrust Misalignment, Pitch: Roll Offset
Thrust Misalignment, Yaw: Rate Gain, Yaw
Thrust Misalignment, Yaw: Dead Band, Yaw
Thrust Misalignment, Yaw: Roll Offset

Third Stage:

Thrust Misalignment, Pitch: Rate Gain, Pitch Thrust Misalignment, Pitch: Dead Band, Pitch

Thrust Misalignment, Pitch: Roll Offset
Thrust Misalignment, Yaw: Rate Gain, Yaw
Thrust Misalignment, Yaw: Dead Band, Yaw
Thrust Misalignment, Yaw: Roll Offset

It can be assumed that an output variable is related to two independent variables by the relation

$$\delta_{W} = f(\delta_{x}, \delta_{y}) = C_{x}\delta_{x} + C_{y}\delta_{y} + C_{xx}\delta_{x}^{2} + C_{yy}\delta_{y}^{2} + C_{xy}\delta_{x}\delta_{y}$$
 (8)

$$\delta w = \underbrace{C_x \delta x + C_{xx} \delta x^2}_{f_1(x)} + \underbrace{C_y \delta y + C_{yy} \delta y^2}_{f_2(y)} + C_{xy} \delta x \delta y \tag{9}$$

$$C_{xy} = \frac{\delta w - f_1(x) - f_2(y)}{\delta x \delta y}$$
 (10)

In this equation $f_1(x)$ and $f_2(y)$ are represented by the piecewise linear functions described in the previous paragraph.

The effect of carrying out the three-step procedure outlined above has to reduce the nonlinear process (i.e., the system and its environment) to a polynomial approximation. This procedure reduced machine running time per sample from 2 to 3 minutes on the TRW Systems N-Stage program to 2 seconds on the Monte Carlo Error Analysis Program, while still including all significant non-linearities.

This polynomial fit was used to relate the input quantities to the state vector at burnout, and any unpowered propagations to other points on the trajectory were calculated by an analytical ephemeris generator.

Monte Carlo Analysis

Using the nomenclature of the previous section, let us assume that there are q error sources, of which p are found to have either nonlinear, cross-coupling terms or both. Since the nonlinear terms have $\pm l\sigma$, $\pm 2\sigma$ and $\pm 3\sigma$ deviations, each nonlinear error source will have a 6x6 matrix of sensitivity coefficients, i.e., the value of the output state vector as a function of $\pm n_{\sigma}$. These values will be used to construct a piece-wise linear function as seen in Figure 3 and replace the square terms $C_{ijk}^{(2)} \delta P_j^2$ in Eq. 7. The advantage of this approach is that it is not necessary to fit an arbitrarily truncated power series when the curve may contain higher components.

For each cross coupling pair of error sources, there will be a single coefficient for each output state vector component or a 6×1 vector for the output state vector. The error sources which have only linear terms can all be compressed into a single equivalent variable by the following steps. Given that:

$$X = [C] P$$
 (11)

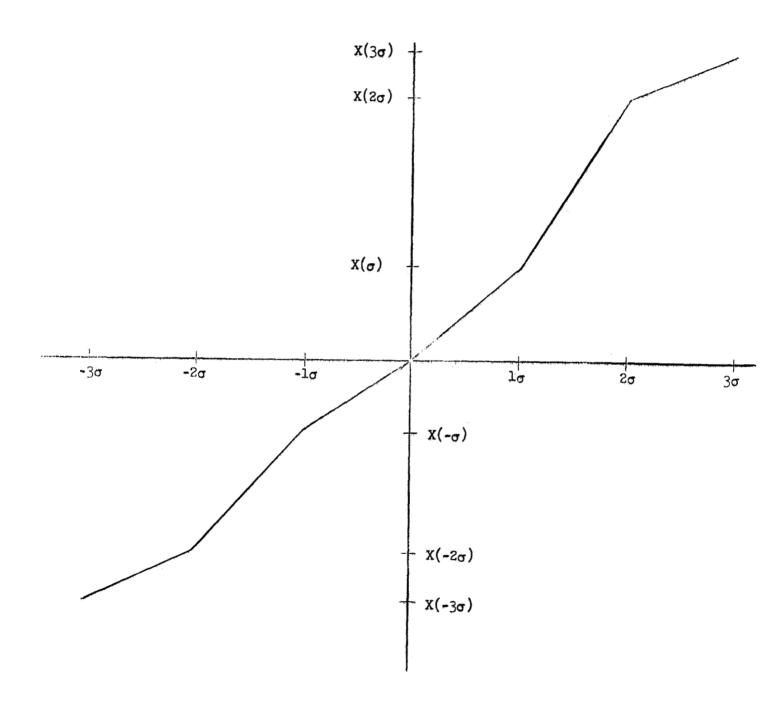


Figure 2. Piecewise Nonlinear Function

where

X is the output 6xl state vector.

P is the linear error source vector of dimension q-p,

C is the 6 x (q-p) matrix of sensitivity coefficients, then $XX^{T} = CPF^{T}C^{T}$.

Taking the expectation of each side gives

$$\Sigma_{x} = C \Sigma_{p} C^{T}$$
 (12)

where

 $\Sigma_{\mathbf{y}}$ is the covariance matrix of the output state vector due to linear error sources;

 Σ_{D} is the covariance matrix of the linear error sources.

Thus, all the linear error sources and their statistics can be compressed into a single 6x6 covariance matrix.

The Monte Carlo analysis progresses through the following steps:

- (1) Using the equivalent linear covariance matrix and a random vector generator, choose a random output vector from this covariance matrix $\Sigma_{\mathbf{v}}$.
- (2) Using the statistics of each nonlinear source, choose a random value of this error source. Enter the piecewise linear function associated with the source and calculate the value of the output state vector dispersion due to this nonlinear effect.
- (3) Using the statistics of each pair of cross coupling error sources choose pair random, values. Multiply the product of these values by the cross coupling sensitivity coefficient vector to calculate the dispersion due to the combined effects of these sources.
- (4) Add the three dispersions due to linear, nonlinear and cross coupling effects to the nominal state vector at burnout to calculate one sample of the analysis. Using this perturbed vector calculate the variables of interest. For the Scout analysis, the following variables were calculated and compared with their nominal values:
 - a. Geocentric Radius Vector
 - b. Inertial Velocity

- c. Inertial Flight Path Angle
- d. Relative Velocity
- e. Flight Path Angle w/r to the Air Mass
- f. Semi-major Axis
- g. Eccentricity
- h. Inclination
- i. Longitude of the Ascending Node
- j. Argument of Perigee
- k. Downrange Distance from Launch Site
- 1. Apogee Altitude
- m. Perigee Altitude
- n. Period
- o. Longitude
- p. Latitude
- q. Altitude

Statistical Analysis of Samples

The analysis of the sample statistics is relatively straightforward. The samples of any variable are used to calculate a mean and second central moment. The samples are then ordered by size and a cumulative distribution is plotted. This curve shows the correct percentile levels for the variable even though the distribution may be highly non-Gaussian.

The equations which are used in the Statistical Processor program (PROC) all be presented here. The same equations apply to both scalars and vectors unless a distinction is made in the description of the sample statistic. All computations are performed in double precision.

AVERAGE OR MEAN VALUE

The sample mean of a random vector X, denoted by X, is computed by PROC using the following equation:

$$\bar{\mathbf{x}} = \frac{1}{N} \sum_{i=1}^{N} \mathbf{x}_{i}$$

In this and all following equations, N is the total number of samples used in the computation and X_i is the value of the random vector X for the i^{th} sample of the Monte Carlo simulation.

COVARIANCE MATRIX AND CORRELATION MATRIX

The covariance matrix of a random vector X, denoted by Σ_{XX} , is defined by the following equation in which E indicates the expectation operator.

$$\Sigma_{XX} = E(X - \overline{X})(X - \overline{X})^{T}$$

This equation can be expanded and rewritten as follows:

$$\Sigma_{XX} = \mathbb{E}(XX^{T}) - (\overline{X})(\overline{X})^{T}$$

The equation used by PROC to compute a sample covariance matrix is similar and can be written

$$\Sigma_{XX} = \frac{1}{N-1} \begin{bmatrix} N & & & \\ \Sigma & X_{1}X_{1}^{T} & - & \frac{1}{N} & \begin{pmatrix} N & & \\ \Sigma & X_{1} \end{pmatrix} & \begin{pmatrix} N & & \\ \Sigma & X_{1} \end{pmatrix}^{T} \end{bmatrix}$$

CUMULATIVE DISTRIBUTION FUNCTION

A cumulative distribution function (CDF) is the probability that a random variable, X, is less than a specified number. This information is displayed by PROC as a graph of the probability that the random variable is less than the specified number versus the specified number. This is accomplished by ordering all of the samples in the order of increasing numerical value and plotting the percentage of the samples that are less than a given value, say ν , as a function of ν .

In Table 1 are listed all the error sources and their magnitudes as they were listed in Reference 1. The items in part I of the table were not the subject of any investigation in Phase I of the contract. Those in part II were the results of an investigation of log book and test data performed by TRW and described in detail in Reference 1. The fifth stage error used in the escape mission were supplied by the Scout Office in a letter.

I. Errors From Previous Investigations

Descri	ption	<u>Symbol</u>	lo Mag. <u>Mag.</u> *
IA Fi	rst Stage Errors		
1.	Propellant Weight Uncertainty	PWLO	.0006
2.	First Stage Inert Weight	SIWL	.0083
3.	Second Stage Inert Weight	SIW2	.0041
4.	Third Stage Inert Weight	SIW3	.00093
5.	Fourth Stage Inert Weight	SIW3	.0024
6.	Specific Impulse	ISPl	.0018
7.	Mass Flow Rate	MFR1	.014
8.	Thrust Misalignment - Pitch	TMPl	1.67 mrad
9.	Thurst Misalignment - Yaw	TMYl	1.67 mrad
10.	Drag Coefficient (CAO)	CAOL	.01
11.	Normal Force Coefficients (CNa)	CNAL	.2
12.	Normal Force Coefficients (CN&q)	CNDQ	.2
13.	Side Force Coefficients (Cyß)	CYBA	•33
14.	Side Force Coefficients (Cyor)	CYDR	•33
15.	Roll Moment Coefficients (Clop)	CLDP	.1
16.	Roll Moment Coefficients (Clp)	CLP1	.1
17.	Pitch Moment Coefficients (Cmo	CMOl	.05**
18.	Pitch Moment Damping Coefficients (Cmδq)	CMDQ	.002
19.	Pitch Moment Coefficients (Cmq)	CMQl	.002
20.	Pitch Moment Coefficients (Cma)	CMAL	.05
21.	Yaw Moment Coefficients (Cnβ)	CNBA	•33
22.	Yaw Moment Coefficients (Cnor)	CNDR	•33
23.	Yaw Moment Damping Coefficients (Cnr)	CNRL	•33
24.	Density Variation	DRHO	.0667
25.	Wind Profile	FWNl	-
26.	Jet Vane Drag Coefficient	CDVl	.1
27.	. Jet Vane Drag Coefficient	CDV2	.1
28.	Jet Vane Drag Coefficient	CDV3	.1
29.	. Jet Vane Side Force Coefficient	LDA2	.1

^{*} Where no units appear, the magnitude is a fraction of nominal.

^{**} This error source magnitude was inadvertently input to both analyses as .002. The effect of the error source was so small, however, that it could not effect the results even if the lo value were increased 25 times.

Descrip	tion	Symbol	lo Mag. Mag.
30.	Roll Moment due to Yaw Axis Shift of Static		
	Margin	LSMY	.01
31.	Roll Moment due to Pitch Axis Shift of Static		
	Margin	LSMP	.01
32.	Pitch Moment due to Roll Axis Shift of Static		
	Margin	MSMR	.1
33.	Pitch Moment due to Yaw Axis Shift of Static		
	Margin	MSMY	.1
34.	Yaw Moment due to Pitch Axis Shift of Static		
	Margin	NSMP	.1
35,	Yaw Moment due to Roll Axis Shift of Static		
	Margin	NSMR	.1
IB Sec	ond Stage Errors		
36.	Propellant Wt Uncertainty	PW20	.00054
37.	Specific Impulse	ISP2	.00094
38.	Mass Flow Rate	MFR2	.Ol
39.	Thrust Misalignment - Pitch	TMP2	1.67 mrad
40.	Thrust Misalignment - Yaw	TMY2	1.67 mrad
41.	Second Stage Aerodynamics (CDO)	CDO2	.1
42.	Second Stage Aerodynamics (Cna)	CNA2	.1
43.	Second Stage Aerodynamics (§)	ZET2	.1
IC Thi	rd Stage Errors		
44.	Propellant Weight Uncertainty	PW30	.0006
45.	Specific Impulse	ISP3	.0014
46.	Mass Flow Rate	MFR3	.018
47.	Thrust Misalignment - Pitch	TMP3	.557 mrad
48.	Thrust Misalignment - Yaw	TMY3	.557 mrad
ID Fou	rth Stage Errors		
49.	Propellant Weight Uncertainty	PW40	.00034
50.	Specific Impulse	ISP4	•006

<u>Description</u>	<u>Symbol</u>	lo Mag. Mag.
51. Mass Flow Rate	MFR4	.018
52. Thrust Misalignment - Pitch	TMP1	.5 mrad
53. Thrust Misalignment - Yaw	TMYl	.5 mrad
54. Coning Rate - Pitch	W4CP	.03 rad/sec
55. Coning Rate - Yaw	MACX	.03 rad/sec
ET. Error Sources Resulting from Investigation		
	Symbol	lo Mag.
IIA First Stage Errors		
56. Proportional Gain Error - Pitch	KPP1	.0243

			Symbor	to mag.
II <i>I</i>	A Fir	rst Stage Errors		
	56.	Proportional Gain Error - Pitch	KPP1	.0243
	57.	Proportional Gain Error - Yaw	KPY1	.0306
	58.	Proportional Gain Error - Roll	KPR1	.0233
	59.	Rate Gain Errors - Pitch	KRP1	• 044
	60.	Rate Gain Errors - Yaw	KRYl	.021
	61.	Rate Gain Errors - Roll	KRRl	.0262
	62.	Random Uncompensated Gyro Drift - Pitch	DTEP	.125x10 ⁻⁵ rad/sec
	63.	Random Uncompensated Gyro Drift - Yaw	DTEY	.200×10 ⁻⁵ rad/sec
	64.	Random Uncompensated Gyro Drift - Roll	DTER	.328×10 ⁻⁵ rad/sec
	65.	Mass Unbalance Roll Gyro - Input Axis	KRIA	1.03 ×10 ⁻⁷ rad/sec/ ft/sec ²
	66.	Mass Unbalance Pitch Gyro - Spin Axis	KPSA	1.08 ×10 ⁻⁷ rad/sec/ ft/sec ²
	67.	Mass Unbalance Yaw Gyro - Input Axis	KYIA	1.03 ×10 ⁻⁷ rad/sec/ ft/sec ²
	68.	Anisoelastic Errors - Pitch	KPAN	3.13×10 ⁻¹¹ rad/sec/ ft ² /sec ⁴
	69.	Anisoelastic Errors - Roll	KYAN	3.13×10 ⁻¹¹ rad/sec/ ft ² /sec ⁴
	70.	Anisoelastic Errors - Yaw	KRAN	3.13×10 ⁻¹¹ rad/sec/ ft ² /sec ⁴

		Symbol	lo Mag.
71.	Rate Gyro Bias - Pitch	DPBE	3.57 mrad/sec
72.	Rate Gyro Bias - Yaw	DYBE	3.57 mrad/sec
73.	Rate Gyro Bias - Roll	DRBE	3.57 mrad/sec
74.	Rate Gyro Misalignment - Yaw	TYRG	1.45 mrad
75.	Rate Gyro Misalignment - Roll	TRRG	1.45 mrad
76.	Vehicle Alignment Errors - Pitch	THOP	5.76×10^{-5} rad
77.	Vehicle Alignment Errors - Yaw ;	THOY	5.43×10^{-5} rad
78.	Vehicle Alignment Errors - Roll	THOR	5.25×10^{-5} rad
79•	Intervalometer and Torquer Scale Factor	DKSG	.0035
Fin	Misalignment Errors		
80.	Normal Force Error/Yaw Fins	CNDR	$.576 \times 10^{-4}$ rad
81.	Side Force Error/Pitch Fins	CYDQ	.576x10 ⁻⁴ rad
82.	Roll Moment Error/Yaw Fins	CLDR	$.576 \times 10^{-4}$ rad
83.	Roll Moment Error/Pitch Fins	CLDQ	.576×10 ⁻⁴ rad
84.	Pitch Moment Error/Yaw Fins	CMDR	$.576 \times 10^{-4}$ rad
85.	Yaw Moment Error/Pitch Fins	NCDQ	.576x10 ⁻⁴ rad
86.	Timer Error - First Step	TIML	.078 sec.
87.	Timer Error - Second Step	TIM2	.004 sec.
88.	Timer Error - Third Step	TIM3	.003 sec.
89.	Timer Error - Fourth Step	TIM4	.003 sec.
IIB. S	econd Stage Errors		
90.	Rate Gain Error - Pitch	KRP2	.044
91.	Rate Gain Error - Yaw	KRY2	.026
92.	Dead Band Error - Pitch	DBP2	.1
93.	Dead Band Error - Yaw	DBY2	.1
94.	Roll Offset	ROE2	.25 deg.
95.	Control Motor Misalignment - Pitch axis	C2PY	.0033 deg.
96.	Control Motor Misalignment - Yaw axis	C2YP	.0033 deg.
97•	Timer Error - Sixth Step	TIM6	.003 s ec.
98.	Timer Error - Seventh Step	TIM7	.003 sec.

IIC. Third Stage Errors

		Symbol	lo Mag.
99•	Rate Gain Error - Pitch	KRP3	.044
100.	Rate Gain Error - Yaw	KRY3	.021
101.	Dead Zone Error - Pitch	DBP3	.1
102.	Dead Zone Error - Yaw	DBY3	.1
103.	Roll Offset	ROE3	.25 deg.

IID. Fifth Stage Errors

		Symbol	lo Mag.
104.	Mass Flow Rate	MFR5	.018
105.	Propellant Weight Uncertainty	PW50	•00033
106.	Specific Impulse	ISP5	•0015
107.	Fifth Stage Inert Weight	SIW5	•00033
108.	Thrust Misalignment, Pitch	TMP5	.667 mrad
109.	Thrust Misalignment, Yaw	TMY5	.667 mrad
110.	Coning Rate, Pitch	W5CP	.0463 rad/sec
111.	Coning Rate, Yaw	WSCY	.0463 rad/sec

REENTRY MISSION ERROR ANALYSIS RESULTS

The mission input data is received from the Scout Program Office and the nominal trajectory which resulted is presented in Table 2.

Table 3 lists the mean, variances and extreme values of the output parameters and Table 4 is a summary of the most significant error sources and their three sigma contributions to the dispersions in range velocity, flight path angle and inclination.

Tables 5, 6, 7 and 8 are the individual three sigma contributions of each error source to altitude dispersion, velocity, flight path angle and inclination measured at each stage ignition and the nominal 400,000 ft time.

Tables 9 and 10 present the non-linear and cross-term data which went into the Monte Carlo results. In Table 9, there is presented the non-linear data for eight different sources. The position and velocity deviations in a radial, down-range and out-of-plane coordinate system were calculated for an error source input of $+3\sigma$, $+2\sigma$, $+1\sigma$, -1σ , -2σ and -3σ . The Monte Carlo program then interpolated between these points.

Table 10 displays the cross-term data involving the effects of a pair of error sources acting simultaneously with both sources at their 3σ level. The sources chosen were all in the control system which will not respond except in the presence of a disturbing force. Thrust misalignment, being the largest disturbing force was chosen to excite these errors. The deviations are represented in the same radial, downrange and out-of-plane coordinate system and they have been normalized by dividing by nine times the product of the two standard deviations. Analysis of these results by comparing the deviations with the 3σ thrust misalignment deviations indicates no significant cross term effects.

Figures 3-19 depict the cumulative distribution function of each of seventeen output variables.

The results indicate that the velocity dispersion is approximately 15 ft/sec greater than in the polar case examined in Phase I mainly due to increased fourth stage error sensitivities. As an example the specific impulse sensitivity increased from 18.3 ft/sec (3) to 132.5 ft/sec. The coning rate effect also increased from 12.3 ft/sec to 107 ft/sec. This increase is partly due to the fact that the errors have been propagated to 400,000 ft whereas in the polar case, they were depicted at fourth stage burnout. While most of the significant error sources showed some increase, the new fourth stage data and pitch profile has evidently enhanced the

effects of the fourth stage errors. The dispersion in flight path angle and inclination also increased from .262 deg and .258 deg respectively to .553 deg and .432 deg (1σ) due to the increased effect of coning rate errors.

It should be noted that the deviations due to a variation in $C_{m\alpha}$ (CMAL) of .6% (3 σ) was 8 ft and .07 ft/sec. The proper value to use was .15% but an increase in the values by a factor of 25 will still be negligible.

TABLE 2. NOMINAL REENTRY TRAJECTORY PARAMETERS

Time (sec)	Pitch Rate (deg/sec)
0	+.0265
3	-3.00517
8	81401
31	- • 4 75
44	 37069
101	 18193
183	20677
280	15
305	 2

Coast Periods:

First Stage Coast - 2.5 sec

Second Stage Coast - -142.22 sec

Third Stage Coast - 21.00 sec

Launch Azimuth - 127.35 deg

Payload Weight - 476 lbs

Launch Position

Latitude (geodetic) - 37.8479 deg

Longitude - -75.4739 deg

Altitude - O ft

Fourth Stage Data:

% of Burn	CG Location Station (Inches)	I (slug-ft ²)	I _x (slug-ft ²)
0	42.7	131.6	8.62
20	41.2	124.6	8.16
40	39•2	115.5	7.50
60	37.1	105.0	6.57
80	33.8	90.6	5.42
100	29.4	71.5	4.01

TABLE 2. NOMINAL REENTRY TRAJECTORY PARAMETERS (Cont'd)

Spin Rate - 150 rpm

Conditions at 400,000 ft

Time - 506.87 sec

Velocity - 26202 ft/sec

Flight Path Angle - -3.065 deg

Inclination - 50.425 deg

Est. Time of Impact - 861.8 sec

Est. Velocity at Impact - 26332 ft/sec

Est. Latitude of Impact - 6.29 deg

Est. Longitude of Impact - -43.6 deg

TABLE 3. SUMMARY OF THE STATISTICAL DISTRIBUTIONS, OF THE OUTPUT PARAMETERS

DISPERSION OF THE SEMIMAJOR AXIS(FT.)

	DARD DEVIAT		=	-3.9093 1.4404	8410 05	1	
2ND	PERCENTILE	SAMPLE	.=	-7.0428 -3.1352	575E 05	3	i Angar
95TH	PERCENTILE PERCENTILE	SAMPLE		-2.7359 1.9867	975E 05	<u> </u>	
	PERCENTILE EST SAMPLE	SAMPLE	=		050E 05 075E 05		

ECCENTRICITY DISPERSION

HEAN .	= 6.2507092D-04
STANDARD DEVIATION	= 8.48736630-03
SMALLEST SAMPLE	= -3.8723297E-02
2ND PERCENTILE SAMPLE	= -1.6123567E-02
5TH PERCENTILE SAMPLE	= -1.3393990E-02
and the contract of the contra	
95TH PERCENTILE SAMPLE	
98TH PERCENTILE SAMPLE	= (1.9575126E-02
LARGEST SAMPLE	= 2.5485134E-02
INCLINATION DISPERSION (DEG	REES) ~
REAL	* 4.7090175D-02
STANDARD DEVIATION	= 4.32764650-01
SMALLEST SAMPLE	= -1.1757078E 00
2ND PERCENTILE SAMPLE	= -8.0596161E-01
5TH PERCENTILE SAMPLE	= -6.6876984E-01
95TH PERCENTILE SAMPLE	= 6.9594288E-01
98TH PERCENTILE SAMPLE	= 8.3587742E-01
LARGEST SAMPLE	= 3.4188604E 00

LONG. OF ASCENDING NODE DISP	
MEAN - STANDARD DEVIATION	= -7.6104484D-02 = 5.6481545D-01
SMALLEST SAMPLE	= -4.0566330E 00
2ND PERCENTILE SAMPLE 5TH PERCENTILE SAMPLE	= -1.1476784E 00 = -8.8480568E-01
95TH PERCENTILE SAMPLE	= 8.5778236E-01
98TH PERCENTILE SAMPLE LARGEST SAMPLE	= 1.1069622E 00 = 1.5903816E 00
LANGES! SAMPLE	- 1.134030105 00

ARGUMENT OF PERIGEE DIS	SPERSION (DEGREES)
MEAN	= 1.4065755D 00
STANDARD DEVIATION	= 7.5754566D 00
SMALLEST SAMPLE	= -8.4022601E 01
2ND PERCENTILE SAMPI	E = -1.2698379E 01
5TH PERCENTILE SAMPI	E = -1.0115087E 01
95TH PERCENTILE SAMPI	
98TH PERCENTILE SAMPI	<u>F = 1.4435243E 01</u>
LARGEST SAMPLE	= 2.9031933E 01

		RBIT (NM)	
MEAN	=	2.0179964D	00
STANDARD DEVIATION	=	2.2432839D	01
SMALLEST SAMPLE	=	-6.3985839E	01
2ND PERCENTILE SAMPLE	=	-4.6811279E	
5TH PERCENTILE SAMPLE	=	-3.6878784E	01
95TH PERCENTILE SAMPLE	=	3.5184326E	01
98TH PERCENTILE SAMPLE	-	4.1909301F	01
LARGEST SAMPLE	=	1.4571008E	02
RADIUS VECTOR DISPERSION (F	Τ)		
MEAN		-5.5997813D	//2
STANDARD DEVIATION	 _	4.5084073D	
SMALLEST SAMPLE	=	-1.3468700E	
2ND PERCENTILE SAMPLE	=	-1.0193450E	4 12 12 12 12 12 12 12 12 12 12 12 12 12
5TH PERCENTILE SAMPLE			
95TH PERCENTILE SAMPLE	=	6.8894500E	To the second of the second of
98TH PERCENTILE SAMPLE	A de la constitución de la const	8.3900750E	
LARGEST SAMPLE	=	2.8475150E	
		100,721001	,
INERTIAL VELOCITY DISPERSION	(I	FPS)	
MEAN	=	-1.52903770	01
STANDARD DEVIATION	=	8.7046162D	01
SMALLEST SAMPLE	=	-4.6214233E	02
2ND PERCENTILE SAMPLE	=	-1.8284058E	02
5TH PERCENTILE SAMPLE	=	-1.5982788E	02
95TH PERCENTILE SAMPLE	=	1.2659253E	02
98TH PERCENTILE SAMPLE	*	1.4581518E	
LARGEST SAMPLE	=	2.3220508E	02
AIRSPEED DISPERSION (FPS)	•	the specific beautiful beautiful by the same	and the state of t
MEAN	=	-1.3847619D	
STANDARD DEVIATION	=	8.7753732D	
SMALLEST SAMPLE	=	-4.7705200E	
2ND PERCENTILE SAMPLE	=	-1.8989673E	
5TH PERCENTILE SAMPLE		-1.6116284E	
95TH PERCENTILE SAMPLE	=	1.2880640E	A Company Agrangement Comments
98TH PERCENTILE SAMPLE	=	1.4927026E	
LARGEST SAMPLE	=	2.3436499E	
INERTIAL FLIGHT PATH ANGLE DI	SPE	RSION (DEGR	EES)
MEAN		7.3649899D	-02
STANDARD DEVIATION	=	5.5346475D	
SMALLEST SAMPLE	=	-3.8355074E	
2ND PERCENTILE SAMPLE	=	-1.0032183E	
5TH PERCENTILE SAMPLE	=	-7.9017881E	
	=	9.9063351E	* App. 12 (12) 40 (4) (4) 44 (1) (2) (4) (4)
95TH PERCENTILE SAMPLE	The state of the state of the		
95TH PERCENTILE SAMPLE 98TH PERCENTILE SAMPLE	-	1.1529894E	

	ATMOSPHERIC FLIGHT PATH ANGL	E DIS	PERSION	(DEGREES) .
	MEAN		7.63693	3870-02	
,	STANDARD DEVIATION	=		580D-01	,
es:	SMALLEST SAMPLE	=	-3.98130		
	2ND PERCENTILE SAMPLE	=	-1.04099		
	5TH PERCENTILE SAMPLE		-8,20550		
	95TH PERCENTILE SAMPLE	=		264E 00	
1	98TH PERCENTILE SAMPLE	_	TO THE CONTRACTOR OF THE STATE OF	555E 00	100
	LARGEST SAMPLE	*		940E 00	
Sagar Sa	APOGEE DISPERSION (NM)				
				†	
<u> </u>	MEAN	-	-4.5542		
	STANDARD DEVIATION	=	10.10 - 10-10 (10.10 (1	569D 01	
	SMALLEST SAMPLE	=	-1.7564		8
	2ND PERCENTILE SAMPLE	=	-9.0430	7.0	
	5TH PERCENTILE SAMPLE	-	-7.8277	130E 01	
	95TH PERCENTILE SAMPLE	. •	6.7718	170E 01	
	98TH PERCENTILE SAMPLE	310.		732E 01	
	LARGEST SAMPLE	. =	1.3064	270E 02	
	PERIGEE DISPERSION (NM)	,		•	
			1.1.1		
· · · · · · · · · · · · · · · · · · ·	MEAN	#	-8.3138		
	STANDARD DEVIATION	, =		580D 01	
	SMALLEST SAMPLE	=	-1.4029	The second secon	
	2ND PERCENTILE SAMPLE		-8.5427		
	5TH PERCENTILE SAMPLE	•	-6.7454		
	95TH PERCENTILE SAMPLE		33 (2004) S. C. T. S. S. S. T. A. C. S. T. T. T.	967E 01	
·	98TH PERCENTILE SAMPLE	=		331E 01	
	LARGEST SAMPLE	=	1.3261	713E 02	a.
	PERIOD DISPERSION (SEC)				
	MEAN	_	-1.4534	729N 01	
•	STANDARD DEVIATION	=		084D 01	
	SMALLEST SAMPLE	=	-2.6143	· 5	
	2ND PERCENTILE SAMPLE	=	-1.1690		
	5TH PERCENTILE SAMPLE	=	-1.0205		
	95TH PERCENTILE SAMPLE		7.4510		
1.500	98TH PERCENTILE SAMPLE		4 4 4 4 4 4 4 4 4 4 4 4 4 4 4 4 4 4 4 4	785E 01	1.7
	LARGEST SAMPLE	=		169E 02	
		REES)			
	LONGITUDE DISPERSION (DEG		(1985)	New Year	
	LONGITUDE DISPERSION (DEG	_	-3.1089	262D-02	120
	MEAN				
	MEAN STANDARD DEVIATION	=	1.1933	379D-01	:
	MEAN STANDARD DEVIATION SMALLEST SAMPLE	=	1.1933 -7.5856	379D-01 495E-01	
	MEAN STANDARD DEVIATION SMALLEST SAMPLE 2ND PERCENTILE SAMPLE	= = = =	1.1933 -7.5856 -2.8737	379D-01 495E-01 736E-01	:
	MEAN STANDARD DEVIATION SMALLEST SAMPLE 2ND PERCENTILE SAMPLE 5TH PERCENTILE SAMPLE	= = = =	1.1933 -7.5856 -2.8737 -2.1219	379D-01 495E-01 736E-01 063E-01	:
	MEAN STANDARD DEVIATION SMALLEST SAMPLE 2ND PERCENTILE SAMPLE	= = = = =	1.1933 -7.5856 -2.8737 -2.1219 1.6010	379D-01 495E-01 736E-01 063E-01	:

	LATITUDE DISPERSION (DGREE		
	MEAN	#	-8.3753786D-03
	STANDARD DEVIATION	=	1-1141130D-01
	SMALLEST SAMPLE	=	-5.4137802E-01
	2ND PERCENTILE SAMPLE	=	-2.0806718E-01
	5TH PERCENTILE SAMPLE	-	-1.7599368E-01
	95TH PERCENTILE SAMPLE		1.8016553E-01
	98TH PERCENTILE SAMPLE	#	2.2498131E-01
,,,,,,,,,,,,,,,,,,,,,,,,,,,,,,,,,,,,,,,	LARGEST SAMPLE	=	5.1899266E-01
	ALTITUDE DISPERSION (NM)		
			-9.21581240-01
	MEAN		-9.2158124D-01 7.4198997D 00
	MEAN STANDARD DEVIATION		-9.2158124D-01 7.4198997D 00 -2.2166505E 01
	MEAN	• <u>•</u> • • • • • • • • • • • • • • • • •	7.4198997D 00
	MEAN STANDARD DEVIATION SMALLEST SAMPLE		7.4198997D 00 -2.2166505E 01
	MEAN STANDARD DEVIATION SMALLEST SAMPLE 2ND PERCENTILE SAMPLE		7.4198997D 00 -2.2166505E 01 -1.6776343E 01
	MEAN STANDARD DEVIATION SMALLEST SAMPLE 2ND PERCENTILE SAMPLE 5TH PERCENTILE SAMPLE		7.4198997D 00 -2.2166505E 01 -1.6776343E 01 -1.3262297E 01

TABLE 4. SIGNIFICANT ERROR SOURCES AND THEIR THREE SIGMA CONTRIBUTIONS AT REENTRY

Error Source	lo Mag.	Altitude (feet)	Velocity (ft/sec)	Flt. Path Angle (deg)	Inclination (deg)
ISP Specific In	.18% npulse - First	17084 Stage			
MFR1 Flowrate -	1.4% First Stage	-42371	28.2	296	
DKSG Torquer Sca	.35% ale Factor	-6434	25.0		
DPBE Pitch Rate	3.57 mrad/sec Gyro Bias	:	-17.4		
TMP1 Pitch Thrus	1.67 mrad st Misalignment	11222 - First Stage	-113.0		
TMY1 Yaw Thrust	1.67 mrad Misalignment -				
CAO1 Drag coeff:	1% icient - First	-6801 Stage			
DRHO Density	6.67%	-44285	22.2	292	
CDV1 Jet Vane D	10% rag	-4859			
TIM1 First Time:	.078 sec r Switch Uncert	11210 cainty	-48.8		
ISP2 Specific I	.094% mpulse - Second	4102 I Stage	15.2		
MFR2 Flow Rate	1% - Second Stage	13575			
TMP2 Pitch Thru	1.67 mrad st Misalignment	-19234 - Second Stag	49 . 9 e		
TMY2 Yaw Thrust	1.67 mrad Misalignment -	- Second Stage			.216
ISP3 Specific I	.14% mpulse - Third	Stage	33.8		
MFR3 Flow Rate	1.8% - Third Stage		87.1		

TABLE 4. (CONTINUED)

Error Source	lσ Mag.	Altitude (feet)	Velocity (ft/sec)	Flt. Path Angle (deg)	Inclination (deg)
TMP3 Pitch Thrus	.557 mrad st Misalignment		44.0	336	
TMY3 Yaw Thrust	.557 mrad Misalignment -	Third Stage			.254
ISP4 Specific In	.6% npulse - Fourth	Stage	132.5		
MFR4 Flow Rate -	1.8% - Fourth Stage		15.08		
W4CP Coning Rate	.03 rad/sec e - Pitch Axis	11703	29.3	187	1.01
W4CY Coning Rate	.03 rad/sec e - Yaw axis	96393	107.5	1.38	114

TABLE 5. INDIVIDUAL THREE SIGMA ERROR SOURCE CONTRIBUTIONS AT SECOND STAGE IGNITION

6. 3UUUUUE US		2.879395E 1.656464E	つけい	2.738479E 1.686715E
4.100000E-03 9.300000E-04	-2.340000E.02 -7.525000E.01	-5.354370E 00 -1.725281E 00	-3.655432E-02 -1.181409E-02	-5.478239E-03 -1.760054E-03
1.800000E-03	1. 427250E 03	•	2.157663E-01	14.
\$3000E-07	0.0000000000000000000000000000000000000	1.123657E-01	-8.228243E-04	-4.901514E-03 -7.863255E-04
030000E-07		.151123E-		3786E
		.302490E-	9.382972E-04	٠
760000E-05 430000E-05	-2.350000E 01 -5.000000E-01	3.15368/E=01 -1.219482E-01	7.683963E-04	4.862668E-03
280000E-06	•0		-675460E-	.357038E
250000E-06	-1.775000E 01	2.159424E-01	-8.546274E-03	7.5174776-04
• UUUUUUE-Ua	0.3750005	947036E	2020C1C	360880F=0
500000E-03 570000E-03	-6.512000E V4	س د	3068944°	
200	-7.250000E 00	•	-2.301005E-02	
570000E-03	5.830000E 02	89099E 0	*575065E-0	446531E-
450000E-03	•	-2.349854E-02	1.666993E-04 -4.268868F-07	9.93/925E-04 -4.268868F-07
.450000E-05	-1 3635005 03	1 723877E 00	1	-357415E-
2.430000E-02 3.060000E-02	30 300078 3.T.	0.	-4.268868E-07	-4.268868E-07
.330000E-02	•	1.544189E-02	-1.203821E-04	
400000E-02	8.425000E 01	67358E	3.268011E	•
100000E-02	0.	1.220703E-04	-1.807866F-04	
120000E-11	700000	770020E-	N	-7.726651E-05
130000E-11	•0	Ь	07547E-	-1.054410E-04
.000000E-03	-4.722500E 02	04907E	.113389E-	.189349E-
		*830994E	.370076E-0	.5/1/62E-0
.000000E-01	0 (.710083E-0	.367433E-U	•
		.796021E-	-09/1515E-	092130E
300000E-01	9	.071045E-	-37.69.1.C-2.	316666
000000E-03	7.00000E 00	. 790039E-0	06875E-	11718
.000000E-03	.250000E 0	•	1	
000000E-03	1	17.	U6 68 6E-U	1 1053035-02
000000E-03	.250000E 0	.0842	1.014/10E-03	
300000E-01	-2.150000E 00	-2.440222E-01	131 00 74	こうこう

-2.390566E-05 -1.049480E-01 -1.621299E-01 1.835613E-05 1.161132E-04	4.012/36E-05 7.957170E-05 -4.268868E-07 2.732076E-05 1.459526E-03 -9.113180E-03	-1.835613E-05 -1.560399E-02 8.426746E-04 -1.002104E-01 -8.522795E-03 -4.028531E-03	-4.028531E-03 7.161880E-03 -6.823444E-02 2.289821E-03 2.061863E-04 3.671227E-05 -1.811336E-01 -7.446396E-01
-2.347878E-06 -4.551358E-01 -1.716715E-01 -2.006368E-04 1.728892E-05	6.616746E-06 9.348821E-05 -1.920991E-06 4.482312E-06 1.709682E-04 -5.009090E-03	9.873892E-04 1.692382E-01 1.287064E-04 -1.832881E-02 -6.235344E-02 -2.941485E-02	-2.941485E-02 -8.072302E-02 7.302484E-01 -2.574298E-02 -2.373918E-03 -4.292347E-04 1.484227E 00 -2.527712E-01
7.324219E-04 -8.883575E 01 1.199286E 01 5.920410E-03	-7.324219E-04 -2.520752E-02 -6.103516E-04 -4.748535E-02 -6.671753E-01		-4.062744E 00 3.152405E 00 -2.099664E 01 6.785278E-01 5.804443E-02 1.013184E-02 -7.178174E 01 3.813843E 00
0. -2.954000E 03 -5.600000E 01 -5.000000E-01	0. 0. 0. -2.500000E-01 -1.400000E 01		1.940000E 02 1.782500E 02 1.655000E 03 5.900000E 00 7.500000E 00 7.500000E 03 8.280000E 03
01 02 05 05	5. 760000E-05 5. 760000E-05 5. 760000E-05 10.000000E-03 10.000000E-03		1.000000E-01 1.000000E-01 7.800000E-02 4.000000E-03 3.000000E-03 1.670000E-03
42 NCR1 43 DRHO 44 FWN1 45 CNDR 46 CYDQ	4.9 CMOR 50 NCDQ 51 LSMY 10	MSMP NSMY NSMR CDV1 CDV2	59 CDV3 60 LDA2 61 TIM1 62 TIM2 63 TIM3 64 TIM4 65 TMP1 66 TMY1

TABLE 6. INDIVIDUAL THREE SIGMA ERROR SOURCE CONTRIBUTIONS AT THIRD STAGE IGNITION

DEV. EG)	-4.320095E-04 -2.437307E-03	.567618E-		378E-	4165E-	-739564E-		.089184E-	309031E-	.331887E-0	07222E-	-397627E-0	.041604E-0	.224703E-0	.650078E-0	.564334E-	279437E-0	.346793E-D	458	-280660E-0	•146085E-	268868E-0	2.877217E-	.405472E-0	5095E-0	.280660E-0	-244340E-0	•439623E-0	3.7F	1024	-523293F-	-330731F-0	.525000E-0	963679	-3.4150956-06	07	4.913040E-03	.941908E-	-1.024528E-05	
TH ANGLE Deg)	-1.893104E-02	3.252499E-0	1.045003E-0	0-	225E-0	.715495E-0	2.561321E-05	54393E-0	.623202E-0	.815799	5775E-0	228E-0	-2.476851E-03	.188827E-0	.109208E-0	9820E-0	93456E-0	.955390E-	-6.936911E-06		-1.489728E-02	•	.071345	ī	.802477E-0	.261380E-0	-802477E-	•470519E-0	74	048202/6-0	5F-0	740596F-0	38674E-	.603441E-0	.053273E-D	*049590E-0	9.	387E-0	20578E-0	
VELOCITY DEV. (FPS)	808228E	3.562744	.144531E C	.295898E-0	89832E	7382E C	74805E-C	-4.357910E-02	80469E-0	029053E-C	€858398E-0	905273E-0	04492E-0	.835938E-0	071436E 0	.504883E-0	.107056E 0	.50468		•	3.250000E 00		5.859375E-03	041626E 0	٠	.643555E-0	.103516E-0	.544922E-0	2832E 0	-29/363E	126709 E-0	672363F-0	-973877E-	.362793E-0	.367188	56152E-0	.48925		.441406E-0	,
RANGE DEVIATION (FT)	-7.290000E 02	3-710230E 1-207500E	.882500E	.120000E	111	.540200E	5.0000005-01	-50000CE-	25000CE		ш	.1.		0000CE	000E	ш	.500000E	.676750E	-5.000000E-01		-5.78500CE 02		E-0	3.327500E 02	500000E-0	.25000E 0	.500000E-0	.50000E-0	O 1	-1.665000E 02	O JORGE O	SOCOCE O	0	.725000E 0	0	.25000E 0	0	.250000E 0	•0	
SIND.DEV.	-000000E-	4.100000E-03	0	-400000E-	1.800000E-03	00000	1-030000-07	1.6300005-07	50000E	•	.430000E	-300008	.250000E	-0000000-	3.500000E-03	.570000E-	3.570000E-03	3.570000E-03	.450000E-	•	.430000E-	.060000E	.330000E	4.400000E-02	.100000E	.130000E	20000E-1	.130000E-	300000D*	Z-000000E-01	300000E-	-300000E-	-000000E-	-3000000°		00000		. 300000	3.30000E-01	
CODE		2 SIMI 3 SIM2	SIN	S	-	1	8 KRIA		•		100	· · · · · · · · · · · · · · · · · · ·	15 OTEP				4,33						100	26 KRP1	25 100			- 1	31 CAUL	32 CNAL	71	35 CYDR		37 CMD0	A16. 3		40 NCBA		42 NCR1	33

				010010	0-3670767
0	6.6/CUUUE-	0 3000007.	10 20200 T	1013010010-41	366615
TWNF	10-3000001.5.1	-2 25000E 02	0.40E	5-506840F=0	707547E-0
CYDO	-30000E-	2.50000E-0	773F-	-934847E-0	-738444E-
CLOR	.76000E-0	. 500000E-0	.662109E-0	-134434E-0	579481E-0
CLDQ	. 760000E-0	000E 0	-1.147461E-02	-4.258196E-05	3.342524E-04
CMDR	5.760000E-05	•0	•0	-4.268868E-07	-30990
NCDO	.760000E-	.500000	-441406E-	.2016516-0	•
SMY	10.00000E-03	00000CE	-221680E-	9/613E=0	
	• 000000	12.00 CE	*0433115	0-3170777	-1071704.
T S S S S S S S S S S S S S S S S S S S	1.	25000E	.103516E-0	.902261E-0	.40330ZE-
MSMK	1	5/250E	.553111E	32115E-0	-10690F-
NSMY	1	-300000	7344E-	.846285E-0	.521816E-
NSMR	1.000000E-01	0 300	7.673340E-01	-5.236300E-04	-4.181058E-02
CDVI	1,0000005-01	-2.013250E 03	-5.615234E 00	-5.362734E-02	-1.328472E-03
CDV2	1.000000E-01	-9.520000E 02	-2.647705E 00	-2.534176E-02	S
CDV3	1.000000E-01	-9.520000E 02	-2.647705E 00	-2.534176E-02	-6.253892E-04
DA2	1.000000E-01	E O	5.273682E 00	51572E-0	6.202665E-04
TIMI	7.800000E-02	7.697250E 03	-4.161023E 01	2.013069E-01	.32
TIM2	300000C	-2.815000E 02	1.407593E 00	m,	۳,
TIM3	3.000000E-03	-2.600000E 01	1.2634285-01	9	
TIM4.	3.000000E-03	-4.50000E 00	4	2625	
TMPI	9	675E 0	-1.100439E 02	2.993736E-01	-2.470138E-02
TMY1	1.670000E-03	E O	07E 0	-9.227762E-02	-3.077820E-01
PW20	٠	90000E	5.676270E-02	-4.970755E-02	-3.384359E-03
SP2		E 0	.963367E	5475E-0	3.0176635-03
MFR2	10.000000E-03	-5.085500E 03	-2.583081E 01	-2.255455E-01	-1.641849E-02
KRP2	1	250E C	5.737671E 00	-6.760014E-02	1.061668E-03
DBP2	0	250E C	E 0	3880E-0	-913593E-
KRY2		00000E 0	1.865385E 00	-1.973423E-02	-5.488911E-03
DBY2	1.000000E-01	О	-1.561279E-01	2.118901E-02	-8.543457E-02
ROE2	2.500000E-01	DOCE O	-1.711304E 00	7.721315E-03	5.201659E-02
C2PY	3.300000E-03	0.	0.	0.	-4.268868E-07
2YP	3.30000E-03	0E-	-2.441406E-04	-1.067217E-07	4.695755E-06
CD02	1.000000E-01	150	.013672E	5331E-0	3.213604E-03
CNA2	1.000000E-01	.602500E 0	8E 0	-1.757968E-02	-1.551435E-02
ZETZ	1.0000000-01	86000E 0	847656	9174E-0	-6.321340E-03
IM6	3.000000E-03	•0	•	٠	-4.268868E-07
IM7	3.000000E-03	0.	7.080078E-03	1.7075475-06	-3.423632E-04
MP2	70000F-	.476	3.238074E 01	59	5.2E-
MY2	1.670000E-03	750E 0	.341675E	.670587E	.049046E-0

TABLE 7. INDIVIDUAL THREE SIGMA ERROR SOURCE CONTRIBUTIONS AT FOURTH STAGE IGNITION

SEV.	-1.233703E-04 -6.313656F-04	2.446061E-	*129010E-0	-215811E-	05420E-0	.846873E-	.762901E-0	18632E-0	\$997689E-0	-309550°	.860142E-0	76887E-	.667972E-0	62500E-0	*5385C0E-0	16844E-	97259E-0	~493659E-	57076E-	9466	68868E-	1.536793E-04	68	-1.370307E-04	91510E-	4		75481E-0	.177123E-	.716014E-0	.708491E-	.793868E-0	42118E-0	.983939E-0	57217E-	9889	.280660	1981E-0	-317995E-	3444E-	.976415
GLE D	-1.199771E-02 -6.465570E-02	.104772E-0	.756831E-0	356E-	.210616E-0	00459E-0	E-0	.507170E-0	.205653E-0	39623E-0	4157E-0	.927573E-0	71746-0	.781062E-0	.604223E	3161E-0	.488813E-0	.657584E-0	257185-0	*084169E-0	34847E-0	-4.734589E-03	1.467423E-07	2.881486E-06	2,3478375-03	-0-	08E-0	*669075E-0	13959E-0	.936472E-0	F-0	.103841E-0	59307E-	.013554E-0	-351595E-0	52267E-0	.326017E-0	3.034898E-05	.438707E-0	0475E-	00619E-0
VELOCITY DEV. (FPS)		2.586670E	*308105E-0	.395020E-0	3853E 0	.593188E 0	9775395-0	.427246E-0	-0-3609560°	*441406E-0	*669922E-0	2.465820E-0	.391602E-0	.187588E-0	2031	.33C664E 0	.586426E-0	.400391E-0	.662720E	-4.394531E-03	• 0	3.586914E 00	Ģ	-9685547E-	168E 0	•	219E-0	82813E-0	.882813E-	*055420E	08	.122070E	.88378	.789063E-0	*159707E-0	.044922E-0	*489258E-0	-7.934570E-02	0C781E-0	.88C859E-0	.441406E-0
RANGE DEVIATION	-9.962500E 02	73750E 0	2500E 0	SOCCCE 0	5CE C	23605CE C	.25000CE C	240000E 0	.25000CE 0	CE 0	OCE C	CE 0	CE 0	O 30	CE C	SCE C		OCCE O	OE O	9	•0	-7.160000E 02	Ů	•	4.05000CE 02	Ċ	-2.50000CE 00	ů	.0	*3260CE 0	SO GE C	OE C	*100000E C	CE O	00009.		OCCCE O	OCE O		. 50000 CE	•
SING.DEV.	0-3000000°	4.100000E-03	.30000E-C	.400000E-0	1.800000E-03	40	9	CBOCOCE-	30000E-	9.250000E-05	5.760000E-05	.430000E+	100	.250COOE-	2.C00C00E-C6	.500000E-0	Q.	.57000E-C	3.570000E-03	-450000E-	11	-	0-	2.330000E-02	9		.130000E-	, med		-3000000°	00000 E-	, CC	*300000E*	.300		2.000000E-03	2.000000E-03	0000	-300000E-	00	300000
36 36	J	ν ν <u>ι</u> ** ν ν <u>ι</u> *	S	SIS	mes	1	8 KRIA				12 THOP	12.1	100	15 DTEP	16 DTEY			1000	20 DPBE	000	1		24 KPR1	25 KPY1	26 KRP1	27 KRY1						13.00	1			37 CMD0	38 CM01	39 CMAL		41 NCDR	42 NCR1

A FWN1	6.670000E-02 10.C00000E-01 5.74000E-05	-2.431700E 04 * -9.95000E 02 -2.75000F 00	-4.985746E 01 1.269873E 01	-3.221433E-01 -2.518632E-03 -1.803597E-05	-5.079099E-03 -3.630288E-02 0.
46 CYDG	-160000E-0	2.50000CE-0	24219E-0	.601342E-0	500E-
	L	•		34847	49E-0
48 CLD0	. 760000E-C	-2.C0000CE 00	-	.057577E-0	5084
49 CMDR	-160000F-0	ċ	2.441406E-04	.402270E-0	•
50 NCD0	5.760000E-05		350	1.467423E-07 -5.980417F-05	5.549529E-06 2.958326E-04
12	C00000E-0	.035000E 0	97266E-	.844547E-0	.601679
M	C00000E-0	500	77539E-0	1542E-0	61321E-
4	C00000E-	025CE C	64917E 0	.603543E-0	.845165E-
1	1.C00000E-01	9	*882813E-0	.511446E-0	81934E-
2.42	C00000E-0	SOCOCE C	5	.317711E-0	-997660E-
57 CDV1	000000	0	.010986E 0	*443234E+0	.726722E-
ပ	000000	13COCE C	849588*	26804E-0	.750236E-0
	1.C00000E-01	0	1.889648E 0	.626804E-0	.750236E-0
60 LDA2	0	OOOCE C	86279E 0	.180565E-0	.134434
	0	0 30009	.624512E 0	.631096E-0	.025151
62 TIM2	4.000000E-03	2500CE 0	.579346E 0	0.15-0	018
	9	SOCOCE C	.428223E-0	*496354E-0	.549529
サ	• COCCOOE-0	CCOCCE C	.587891E-0	.583697E-0	.280660E-
65 TMP1	0-1	025E C	.158186E 0	.816300E-0	.493180E-
 	1.670000E-03	4225CE 0	.155668E 0	0474E-0	.441494E-
2Md	0-300000+*	.917500E 0	-395264€ 0	61299E-0	47248E-
I SP	*400000E-0	.22550CE 0	788330E 0	530E-0	.462052E-
	10.000000E-03	ပ	26489E 0	17127E-0	.825493E-
KRP	.400000E-0	.465500E 0	.485840E 0	.104270E-0	*152594E-0
089	-C000000-	SCE C	36563E 0	70690E-0	.712830E-
72 KRY2	9	37500E 0	.095947E 0	10090E-0	.878925E-
A80	~000000°	.892500E 0	-1.603271E 00	4F10	116-0
Car.	2.500000E-0	*66500CE 0	74756E 0	.034912E-0	.59811
	-300000E-0	• 0			200000
: ب ا ی	9	1		L.46/423E-U	のななのないし
000		30250	9287	-174895E-	-503495E-
78 CNA2	1.C00000E-01	7500CE 0	.186768E 0	.831746E-0	.863255
7	-3000000°	S H	.157715E O	.443878E-0	.4735
inui.	*0000000-0		٠	۰	68868
81 TIM7	3. CGCCCOE-03	2.500000E-01	1796	2208	24623
S IND	-3000	.38210CE C	.203931E.0	*736029E-0	.265719E-
3 TMY	70000E-0	.5560CGE 0	*373047E-0	.778638E-0	.999832E-
4	00000E-0	20	-5.576904E 00	Ш	.448161
85 ISP3	0000	3000CE 0	∘610327E 0	306815-0	2.8691065-03
₩ 9	1.800000E-02	5750CE C	.956714E 0	0-3512	-7.11.081E-03

1.560314E-02	-1-549999E-04	-2.814941 E-01	-1.25000CE 0	Z.50000E-01	ROES
-4.268868E-07	•0-	•0	ô	1.000000E-01	
-4.268868E-07	•0-	°	ċ	1.000000E-01	89 DPB3
-4.20880AE-U	-0-	•0	•	4.4C0000E-02	¥

TABLE 8. INDIVIDUAL THREE SIGMA ERROR SOURCE CONTRIBUTIONS AT RE-ENTRY

1/ PW10 2 SIW1 3 SIW2					
-		11	(FPS)	(DEG)	(0EG)
	•	36	-2.482910E-01	-1.023037E-02	-6.061793E-05
	-300000E-	.567500E	34424E	836E-0	-3.573043E-04
	.100000E-0	.950000E	- 1	-1.838332E-02	-1.186745E-04
	.300000E-0	.485000E	2.629395E-	-5.901310E-03	
SIM4		-2.735000E 02		-1.831078E-03	.109906E
	-800000E-0	.708425E	.093262E	•	.322194E-
	0000E-0	.237125E		.963465E-0	360257E-
8 KRIA	1		.000977E	.617866E-0	.083137E-
	1.08000E-07		141E	.217571E-0	.391510E-
10 KYIA	.030000E-0		-	-3.663222E-05	215472E-
THOR	.250000E-0	'n.	-1.367188E-02	.810481E-0	-791769E-
North	• 760000E-0	.557500E		9.530248E-05	
	30000E-	. 750000E	.318359E-	.811513E-0	814198E
14 ULEK '1 OT'0	.280000E-0	. 150000E	.812500E	.561321E-0	
- 17	• 250000E=0	•430000E	1 1	-5.546861E-05	•
1	.000000.	.250000E	-1.855469E-02	-6.187191E-05	.532383
	.500000E-0	.434000E	2.504932E QD	-2.303214E-03	3.850519E-04
	.570000E-0	.075000E	1.901855E-01	6.920102E-04	
DYBE		.525000E	.188477E		1
	.5 70000E-0	.816250E	-1.742261E OD	.701206E-0	
	.450000E-	.000000E	.441406E	-8.431015E-06	199552
1,30	.450000E-		.441406E-	.600826E-0	.537736E-
6 17	-430000E-	-8.190006 02	3.7526868 00	6.3651496-04	٠
	-00000e-	•			.537736E-
	.330000E	-500000E-	.464844E-		•623114E-
	-400000E-	.347500E	-2.279297E 00	•	•
	2.100000E-02	.500000E-	.882813E-	-3.201651E-07	-2.134434E-06
31.11.12	.130000E-		.123047E	-2.374558E-06	-8.537736E-07
	.120000E-1		-2.441406E-04	-1.600826E-07	.024528E-
× 3	.130000E-1	2.500000E-01	200000	.520784E-0	
	.670000E-0	.122250E	.130752		.144679E-
	.67000E-0		.169922E-	.629772E-0	-8.775640E-02
- 1	.000000.	.801000E	.416992E	-4.378738E-02	-2.744882E-04
33333	*000000E	.925000E	.082275E	3.632860E-03	2.732076E-05
12.50	.00000CO.		٠	-1.537326E-04	-3.329717E-05
A. A.	6	.200000E	2.905273E-02	1.7120835-04	-1.319507E-03
	0-	000		.942851E	1.818538E-04
	.0000000.	.500000E	-2.343750E-01	-3.655218E-06	-4.695755E-06
39 CMD0	.000000E-0	.575000E	1.098633E-01	4.188827E-06	8.537736E-07
CM01	2.000000E-03	4.00		.337117E-0	-1.280660 <u>E-0</u> 6
CMAL	.000000E-0	.750000E	-7.763672E-02	-7.086321E-05	-2.134434E-06

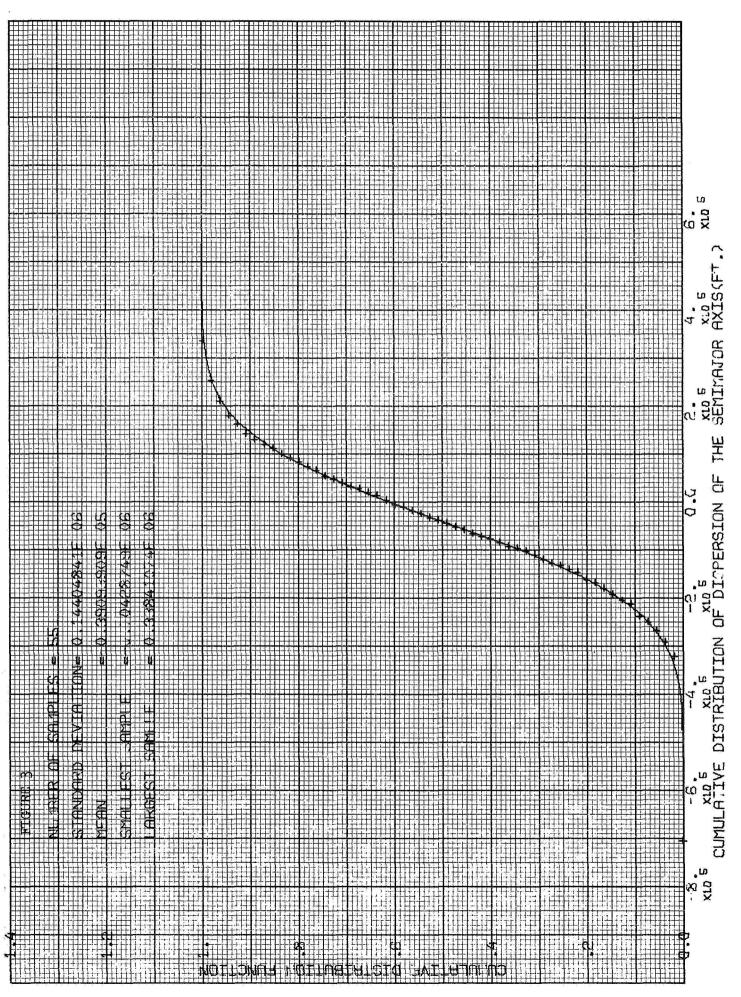
.300000E-01 0670000E-02 -4.428575E .000000E-01 -6.302500E .760000E-05 -3.250000E .760000E-05 -5.000000E .760000E-05 -5.000000E .760000E-05 -7.00000E
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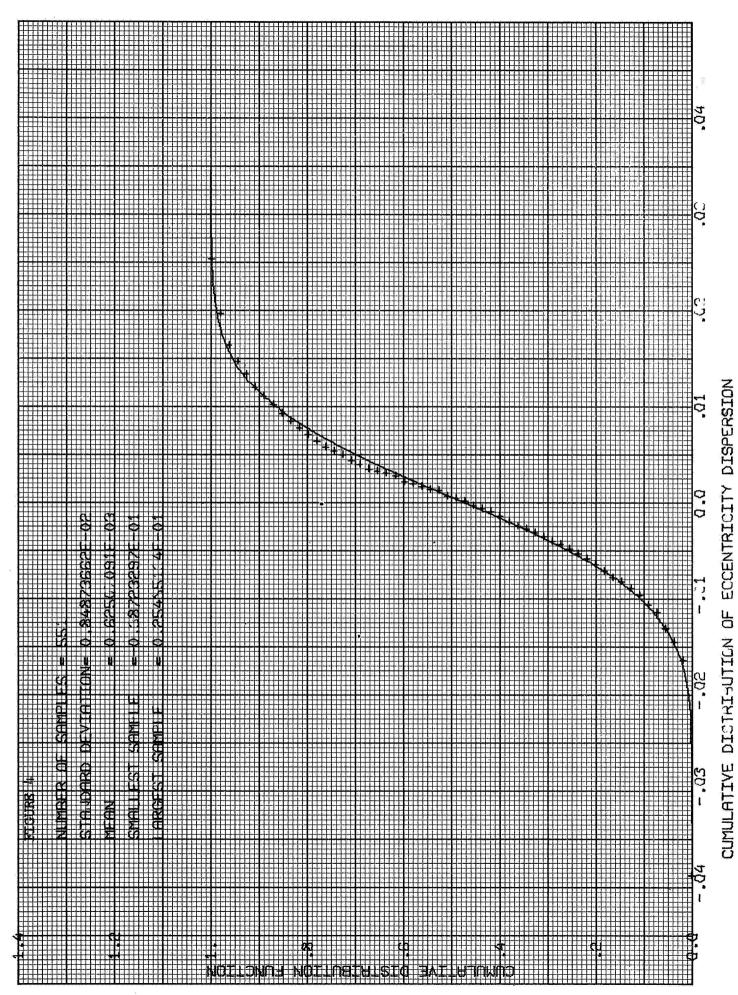
084. pw30	5.00000E-04	-2.707500E 02	-5.211914E.00	-6.538972E-03	-2.309458E-04
85 ISP3	1.400000E-03	1.605250E 03	.388672	4.065854E-02	0
86 MFR3	1.800000E-02	-9.407500E 02	17285	-6.659469E-02	L
87 KRP3	4.400000E-02	•0	•0	1.067217E-07	
88 DPB3	1.000000E-01	*0	•0	1.067217E-07	.537736
89 KRY3	4.400000E-02	•0	• 0	1.067217E-07	36E-
90 DYB3	1.000000E-01	·C	• 0	1.067217E-07	
91 ROE3	2.500000E-01	-1.325000E 01	-1.870117E-01	-2.774764E-04	0
92 TMP3	5.570000E-04	_	4.408228E (D)	-3.365800E-01	E-0
93 TMY3	5.570000E-04	-6.030000E 02	-5.001953E 00	-1.021351E-02	2.542845E-01
05M PW40	3.400000E-04	-2.900000E 01	-5.3999028 00	-1.708561E-03	-1.084293E-04
95 ISP4	6.000000E-03	1.557500E 03	1,325979E (0Z)	9.266979E-02	5.911102E-03
96 MFR4	1.800000E-02	-1.534250E 03	-1.508081E 01	-8.276204E-02	3
97 TMP4	5.000000E-04	-2.305250E 03	1.242188E 00	-3.381037E-02	2E-
98 TMY4	5.000000E-04	-2.745250E 03	O	-3.987310E-02	
99 W4CP	3.000000E-02	-1.170300E OA	-2.936719E (Q1)	-1.878927E-01	1.011932E(00)
100 M4CY	3.000000E-02	9.639325E 04	-1.075078E (02)	1,382834E(00)	1.1443606-01
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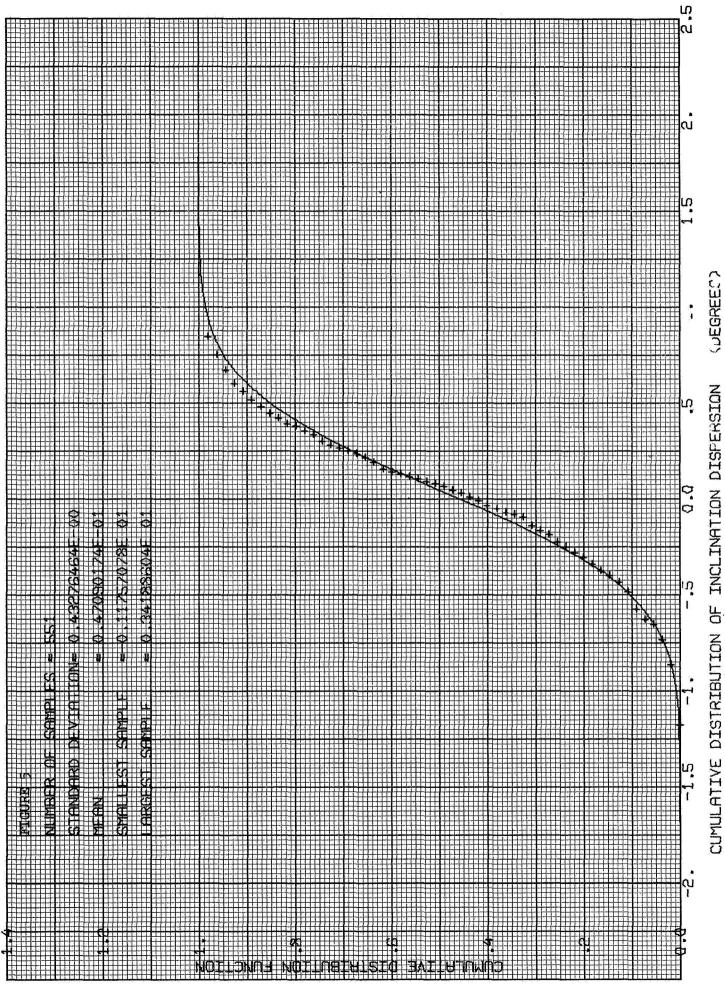
CODE	STND.DEV.		POSITION DE	EVIA	TICNS			VELOCITY D	EVIA	T 1 ONS	
		1.113800E 8.006920E 4.279310E	000900 012630 216790 686920	1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1	1.141500E 7.692140E 3.847650E	03	3.479100E 01 2.316920E 01 1.196220E 01	-1.113900E -7.510120E -3.726400E	0000	2.028500E 0 1.339610E 0 6.672910E-0	00
		1.718020E 2.5776C0E	4 3,366270	000	4.816920	100	4.729150E 0 7.068500E 0	6.250000 9.368600	000	.182680E-0	
MY1	1.6 70000 E-03	COE	3 -6.210903	0	.620500	4	2.058500E 0	-8.241400	0	.293600E 0	أينم
		70020E 35550E	3 -4.142660	00	.082790	4 4	1.372000E 0 6.862950E 0	-5.519260	99	.326690E 0 .769340E 0	
		.312550E .627970E .9366C0E	03 -9.501110E 03 -1.900129E 03 -2.850600E	03 03 03 03 03 03	-1.530010E -3.059340E -4.586600F	444	-8.255510E 00 -1.652370E 01 -2.475100E 01	-1.688820E -3.372090E -4.751500E	000	-2.744880E 01 -5.490070E 01 -8.229500E 01	
.MP2	1.6 70000 E-03			•					(•
		1.9240C0E	1.603800	П 04 24	3.904/00E 2.611360E	020	-6.444400E UI -4.266290E UI	4.654200E 3.022040E	10	7.500440E 00	S 0
		.430120E .066660E	.346270 .899410	00	.300070	2 2	2.133770E 0 3.438550E 0	1,552090	00	.720440E-0	
		122240E	-1.573350	0 0	.785050		.872880E 0	-5.487490	0	.47757CE 0	0
FMY2	1.6 70000E-03	1000001	000106.3)			0 1006 100 6	COOTT 7. 0-	<u>ر</u>	• 1030601	5
		-3.957700E -2.700450E	780510	E 02	-5.871200E -4.019110E	400	-1.302600E 01 -8.671030E 00	2.27280 0E 1.552460E		-1.622900E 02 -1.088420E 02	22-
		4.806620E	1 8.311610	0 0	2.844610	ተጣ	1.617380E-0	2.341080	0	5.04427CE 0	40
3		01130E 48000E	1 1.662010 2 2.499300	00	.520500	ന ന	3.225080E-0 4.846800E-0	4.688570	99	.016790E 0	
ባ ኤ ጀ	100000000000000000000000000000000000000	3.2654COE	7.218300	00	.049800	~ 10	1.651800E 0	3.479300	0	.445000E 0	0 -
		.094560E	2.413090	00	.022690	10	.513920E 0	1.152000	0	. 859980E-0	
		062370E 153770E	22	E 03	-4.993050E -1.009990E	05	2.666760E 01 5.324440E 01	-5.304290E	So	-2.889150E-01 -5.700010E-03	-4 -
[MY3	5.570000E-04	.5174COE	-3.126100	O .	1.478900		.998600E 0	-1.578900	0	7.337800E-0	^
		302100E 203640E 101070E	2 -1:183400 2 -8.001060 2 -3.992960	E 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0	-3.410700E -2.084880E -1.152670E	4400	-3.208500E 0C -2.143090E 0C -1.076770E 0C	0 -5.699100E 0 -3.799000E 0 -1.899420E	000	-1.647000E 02 -1.162340E 01 -5.830100E 00	N=0
4		310040E 274410E	9.083040	CO	.375410	mm -	3.313320E-0	4.020660	00	.114550E 0	0-
.3		.8907C0E	2 2.719303	0	•305200		.707500E-C	1.315600	0	.321300E 0	-

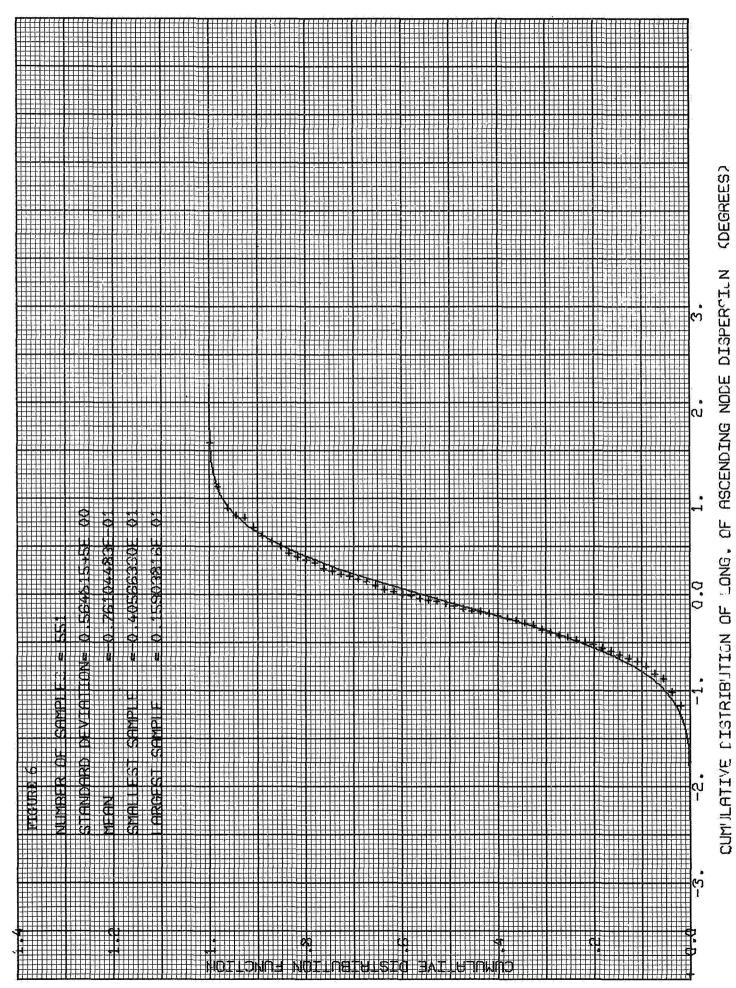
MACP	3.00000 E-02											
		-1.1917COE	04	9009	03	3200	04	-7.913100E			-6.216500E	25
		-7.946580E	03	520E	03	3710	40	.264980E	_		. 1490	20
		-3.973290E	03 -2.118260	2 60 E	03	-3.17614CE	04	-2.637740E	0.1	-1.372210E 01	-2.072220E	.70
		3.995210E	03	3 20 E	03	3.4C5210E	94	.657530E			.2212	0.5
		7.982670E	03	3 C+9	03	6.806270E	9	.319080E	_		.440	02
		1.1985COE	9	3 CO 8	03	400	05	.972600E	, ,i		.6631	02
M4CY	3.C00000E-02											
		9.638700E	04 -1.229	500 E	9	7700	40	6.499300E	~	.093200E	1121006	10
		6.424950E	04 -8-104	270E	03	-7.385010E	03	•	N	30	749010E	01
		3.212940E	04 -4.098130E	130E	03	-3.692440E	03	2.166430E	0.2	.69774	-2.370590E	010
		-3.462090E	9	9 90 E	03	0001	03	•	02	064440E	594110E	01
		-6.925040E	9	050 E	03	8.089010E	03	-4.666020E	02	.412050E	1801108	10
		-1 DAROUP	Ç	A00 F	6	1,212200F	7		2	TO C	782500F	-

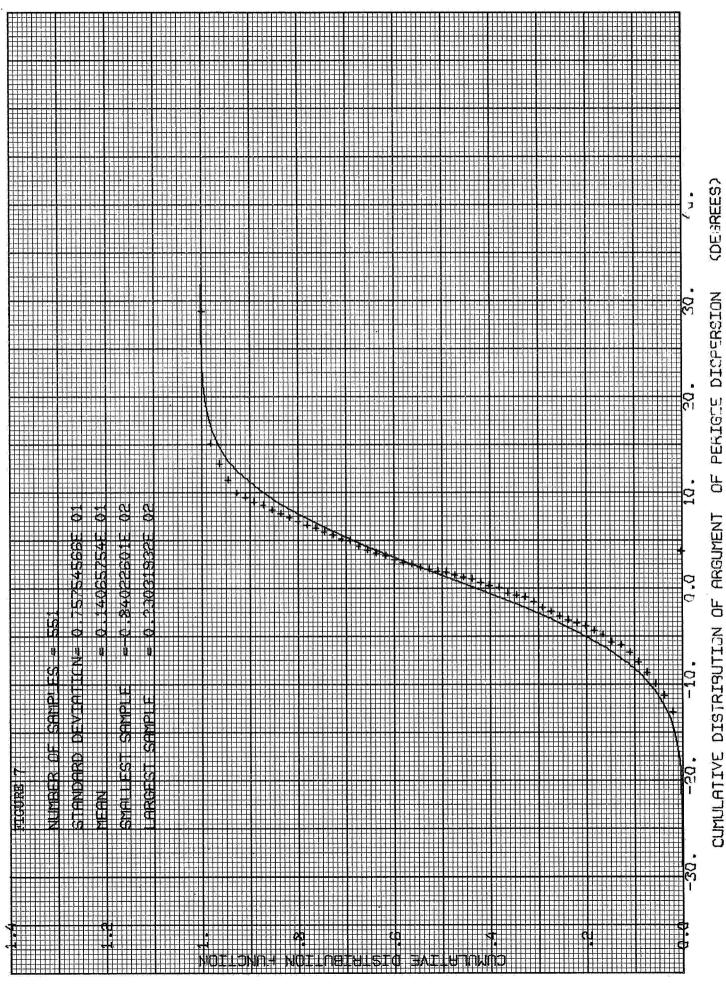
		0		0		0		0		0		Ò		0		0		0		0		0		0
		-1.473889E		2.226114E		-9.675449E		4.285980E		-1.442783E		-4.419861E		-1.482963E		-4.329054E		-8.401587E		-2.254019E		4.384070E		7.652079E
		7 70		40		- 40		- 50				03 -						- 50		- 40		0.4		0.5
		9.735257E 04 -1.473889E		5.836906E 04		2.011619E 04		03 -2:136188E 04 -4.285980E		-2.494017E 03		-1.623342E		-5.363077E 03		-1.495644E 04		-2.752418E 04		3.195147E 04 -2.254019E		9.980111E		1.764244E 05
		C 2		04		5				03		0.2		03		03		04		0.5		CS		0.5
IND.DEV.1 STND.DEV.2 6 CROSS TERMS	670C00E-03 4.400005E-3	5	1.670000E-03 1.000000E-01	1.922848E C7 7.669614E 05 -7.720792E	2.5	6.520423E 06 -3.372101E 05 -2.693384E	1.670000E-03 2.620000E-32	-2.690189E 07 -1.574909E 08 3.320707E	1.6	-9.353201E 05 -5.284571E 07 -5.343820E	2.5	-5.892312E 05 -1.625653E 07 -6.295840E	2.500000E-01 5.570000E-04	-1.117907E 06 -3.070665E 07 -1.017385E	1.0	-3.114164E 06 -8.952605E 07 -8.477071E	5.5	-5.746561E 06 -1.739315E 08 -1.599095E	2.5	95271E 04 -1.530964E	1.000000E-01 5.570000E-04	2.C61374E 07 9.043896E 05 -4.841963E	5.570000E-04	3.631318E 07 1.562803E 06 -8.451101E 05
CODE1 CODE2 STND.DE	KRP	343	2 OP B2	340326E 07	2 TMP2	148860E 06	TMY2 KRY2	064416E 05	2 DYB2	649723E 06	2 THY2	014488E 05	3 TMY3	006235E 05	3 THY3	659506E 06	TMY3 KRY3	131179E 06	3 TMP3	-3.009858E 07	3 TMP3	540070E 07	TMP3 KRP3	659339E 08
000	TMP2	-3.92	A D	4	30	ф	THA	•	¥ E	7	ROE	2	8	-2-	DYB DYB		A ME	Ţ	# 0E3	4	OPB OPB	6-	TE	-1-

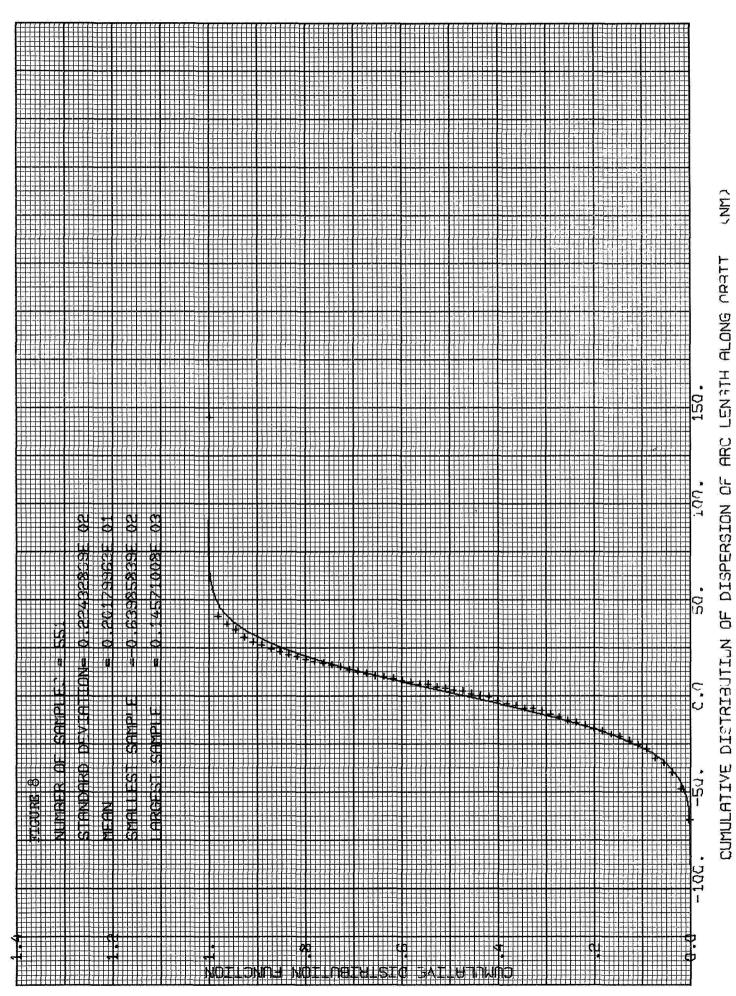


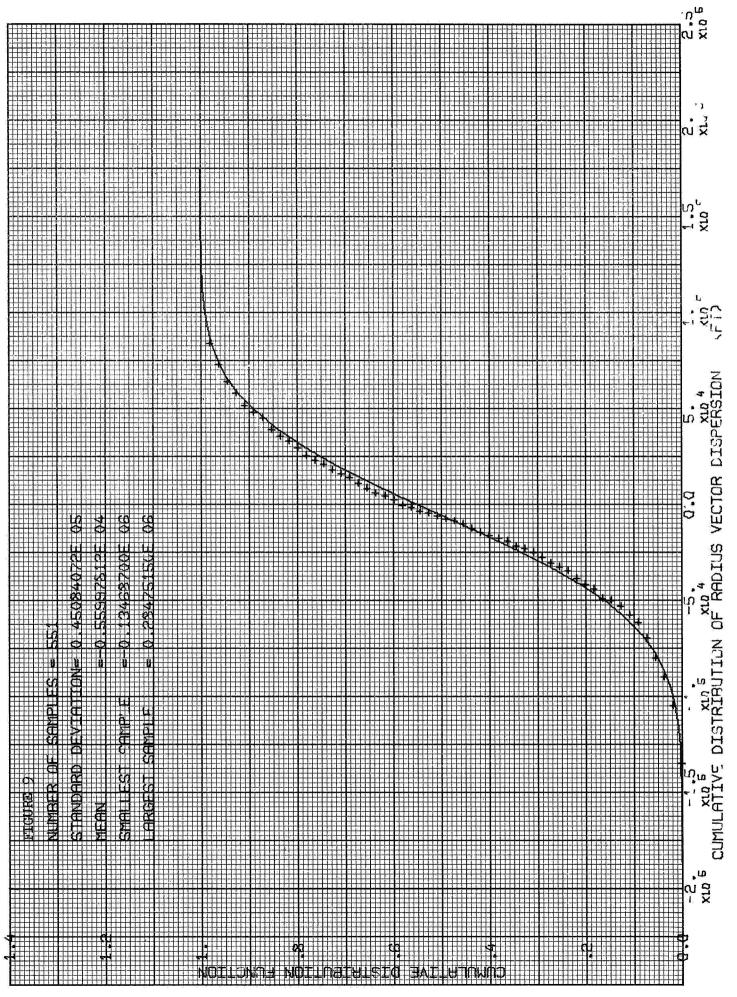


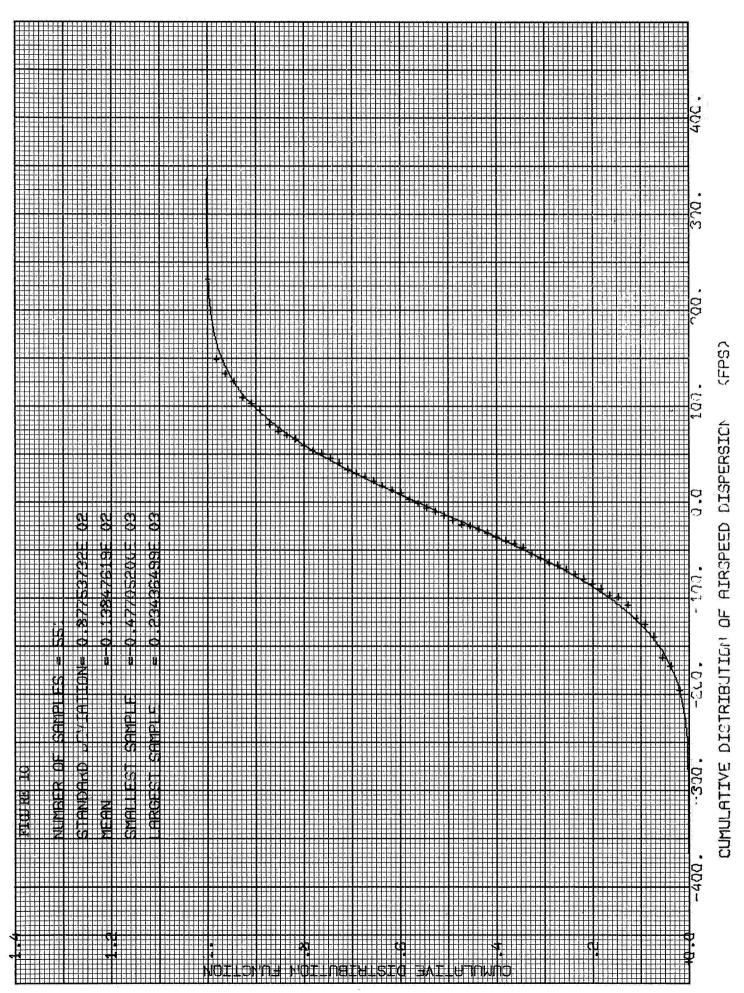


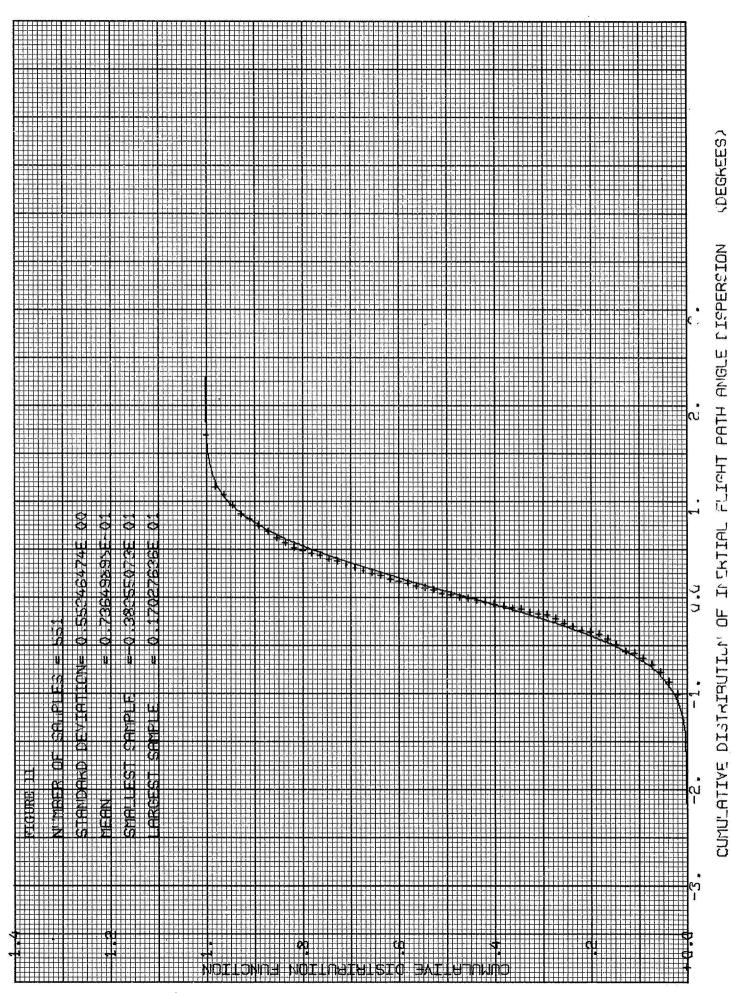


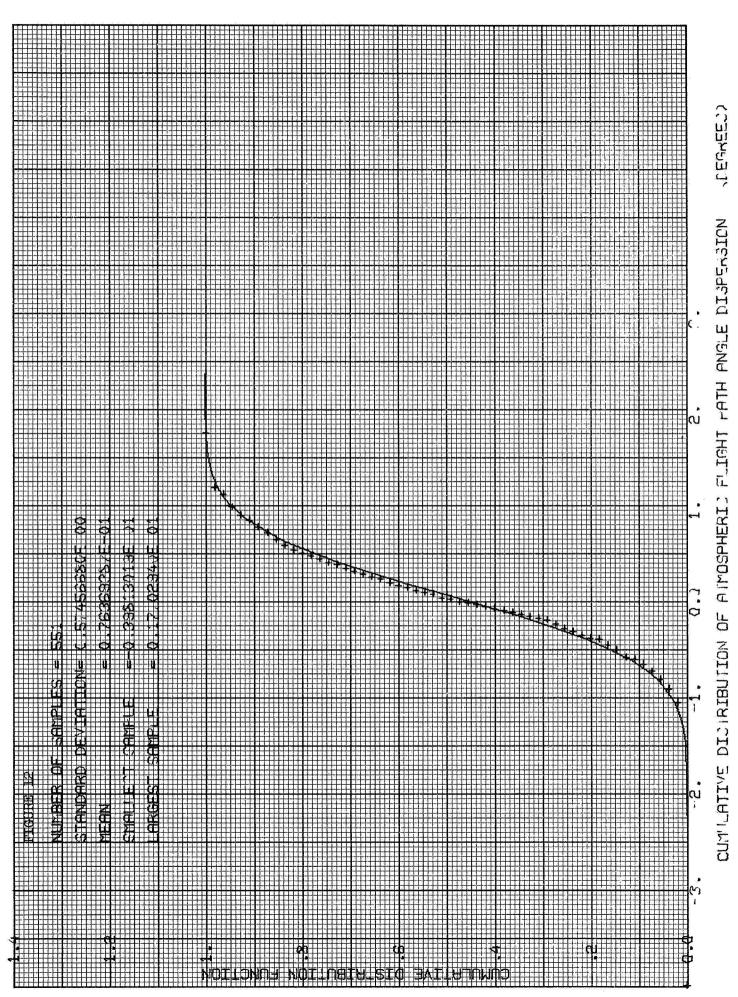


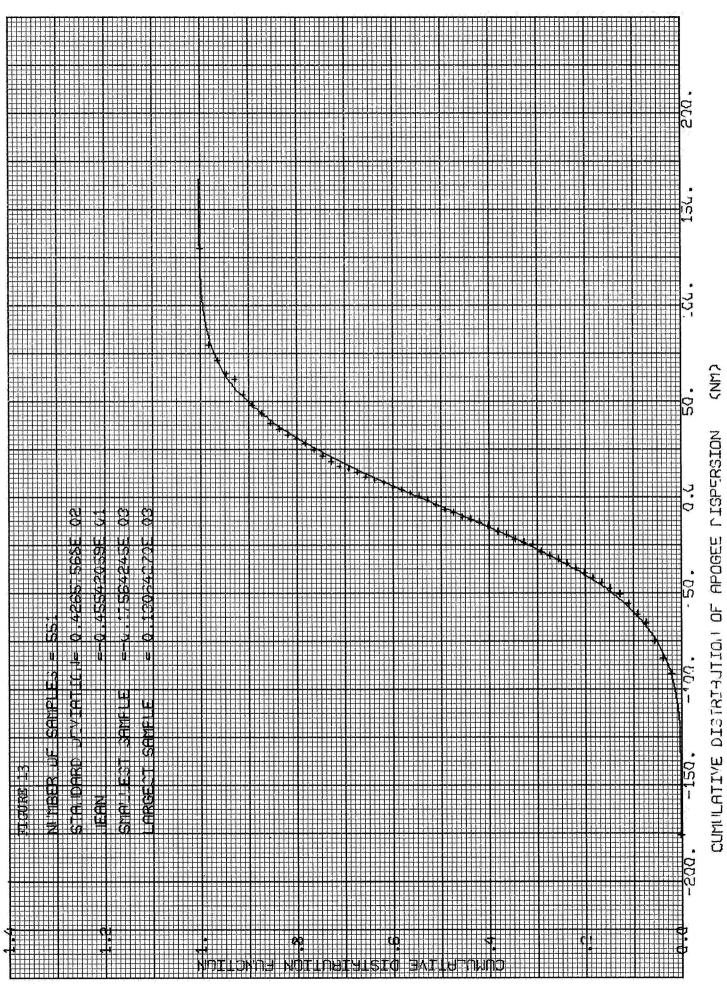


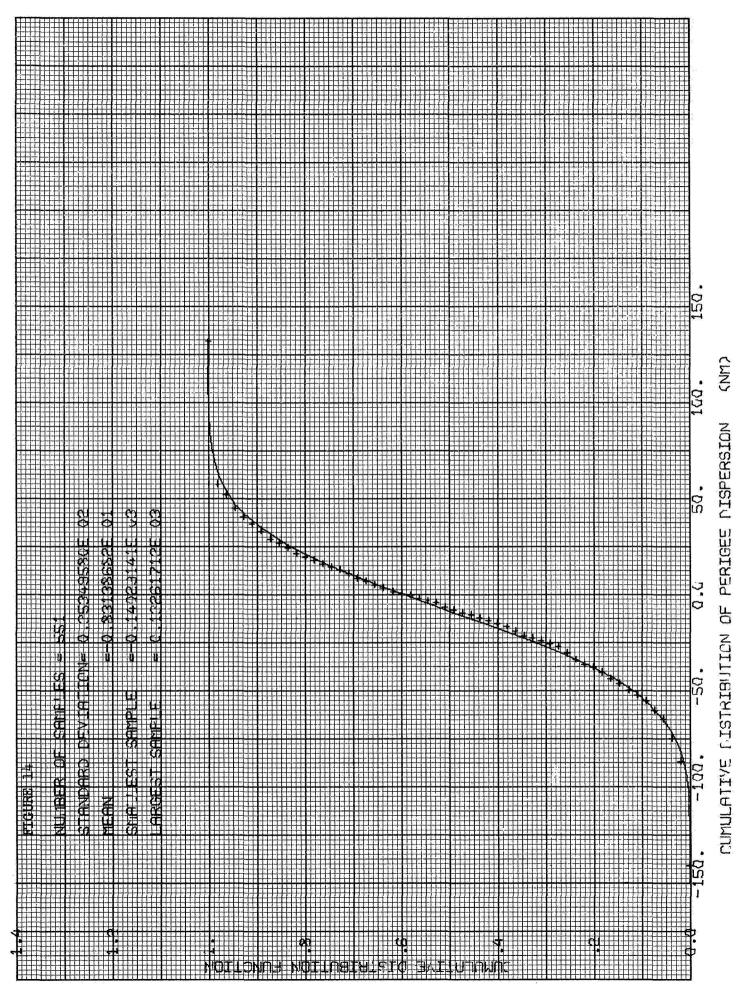


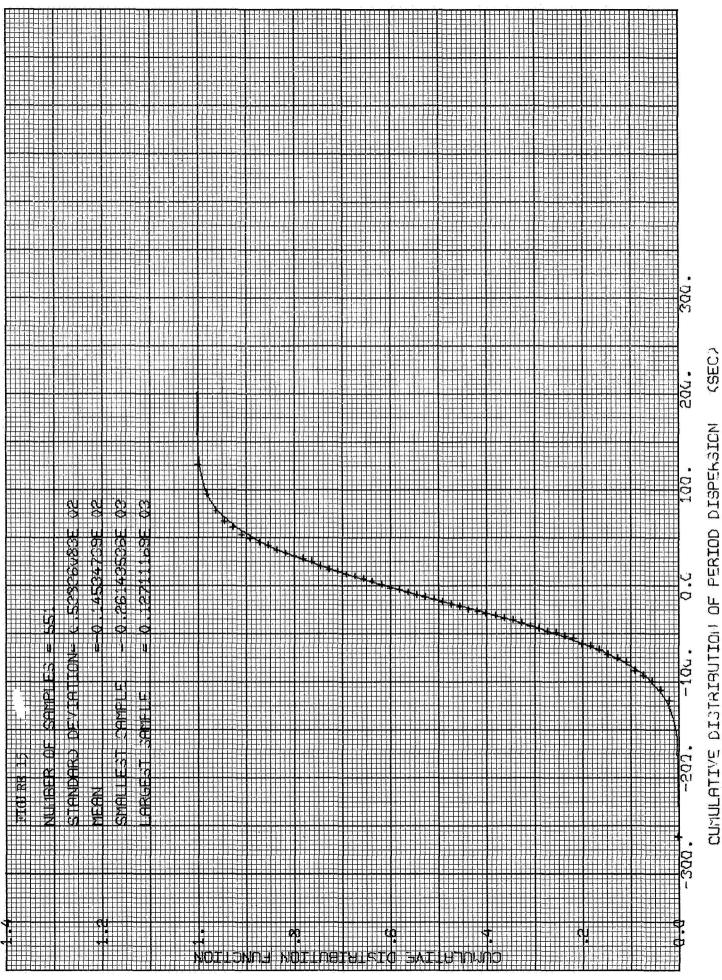


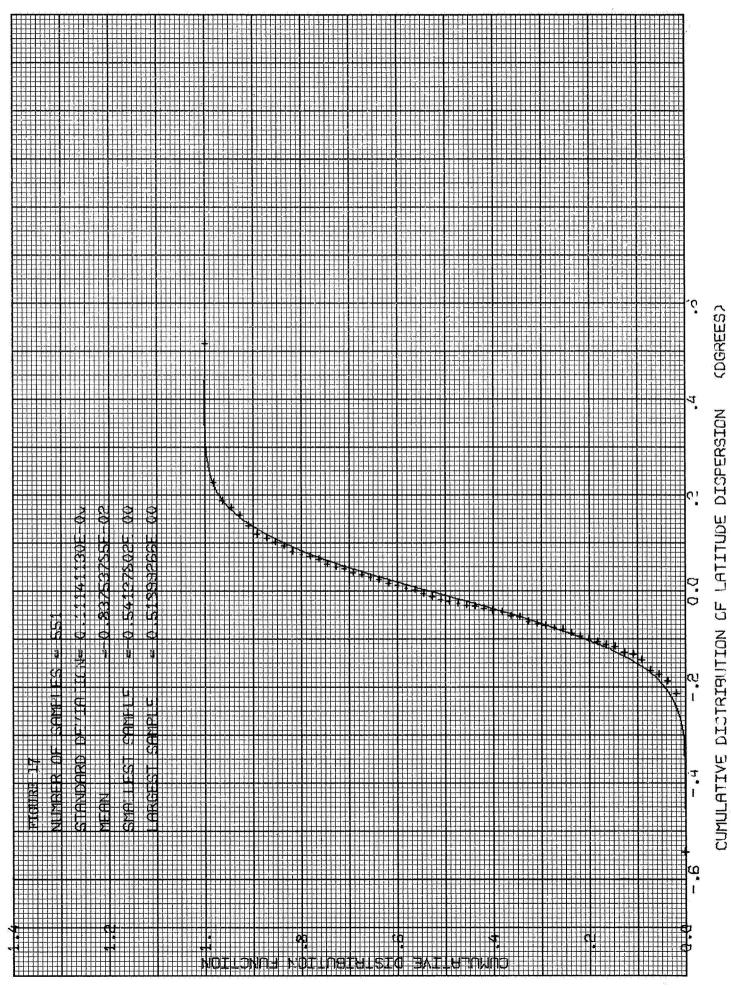


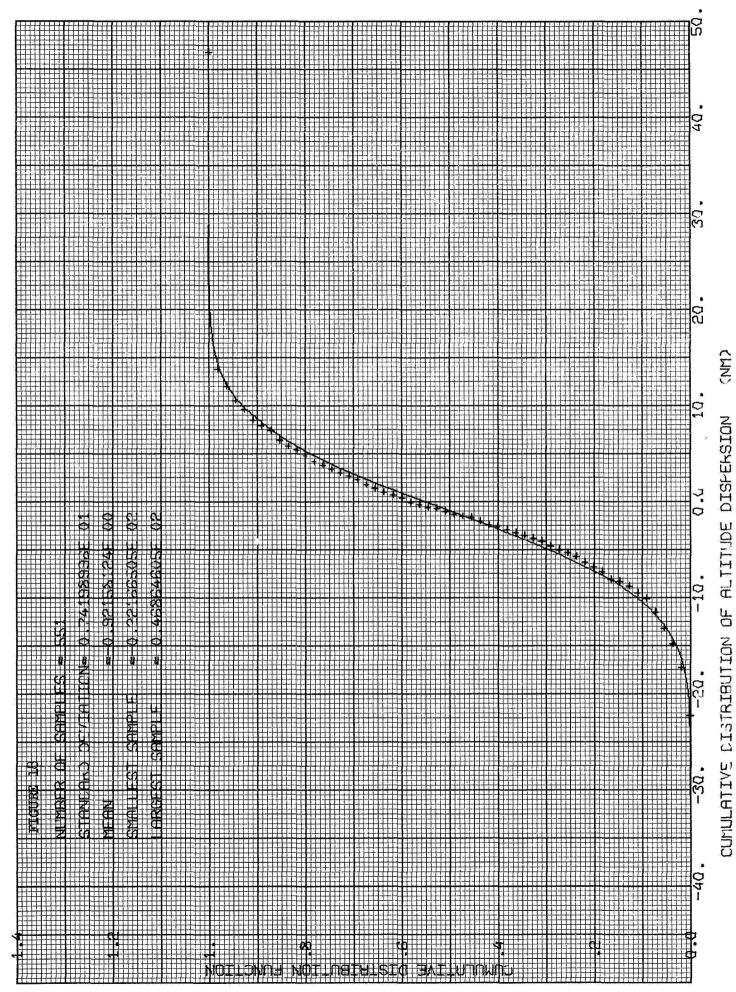


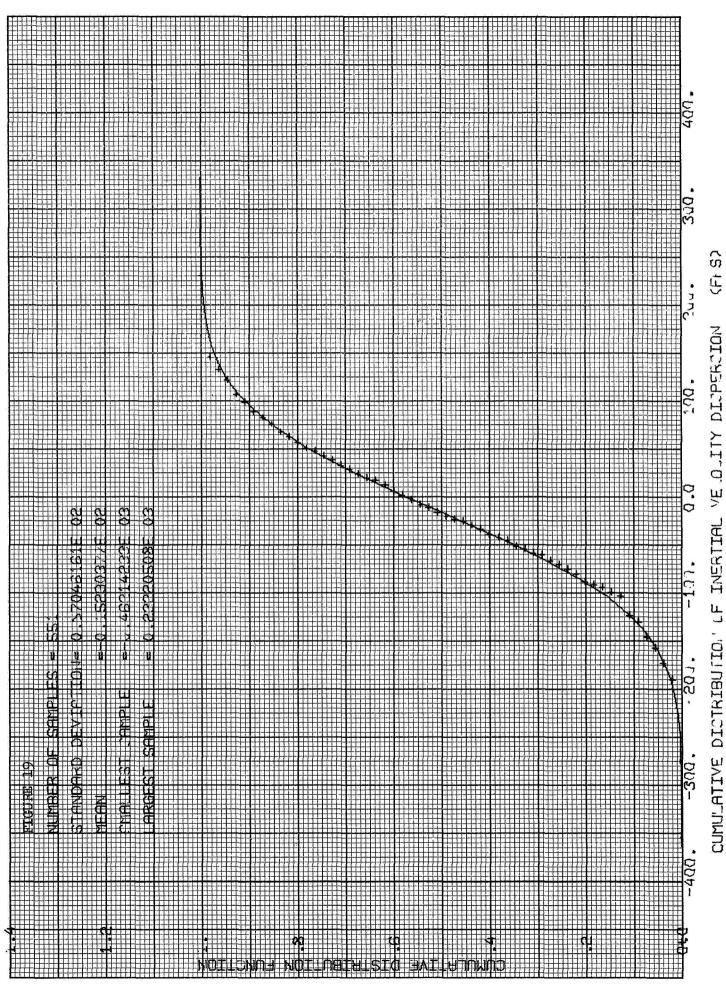












ESCAPE MISSION ERROR ANALYSIS RESULTS

The input data and the resulting nominal trajectory are listed in Table 11. In Table 12 are listed the means, standard deviations and the extreme values for each of the output variables.

Table 13 lists the most significant error sources and their respective three sigma contributions to altitude, velocity, flight path angle and inclination.

Tables 14, 15, 16, 17 and 18 are the individual three sigma contributions of each error source to altitude, velocity, flight path angle and inclination measured at each stage ignition and fifth stage burnout.

Tables 19 and 20 lists the non-linear and cross-term data in this analysis in the same format as previously discussed.

Figures 20-36 depict the cumulative distribution functions for the output variables.

The results of the Monte Carlo analysis show that significant individual contributions are present in all of the stages with no signle source predominating, leading to a total velocity dispersion of 78 ft/sec. The inclination variation is less than in the four stage mission due to the decreased sensitivity caused by a large nominal velocity.

In general, the five burn trajectory was less sensitive to errors in first and second stage than was the four stage mission. This effect compensated for the additional fifth stage errors. The large velocity bias effects present in the four stage mission was also decreased substantially.

TABLE 11. ESCAPE MISSION INPUT DATA AND NOMINAL TRAJECTORY PITCH PROFILE:

Time (sec)	Pitch Rate (deg/sec)
0	0
3•	-2.625
10.	813
31.	700
42.2	461
76.0	081
83.42	- •432
126.44	+ .400
144.83	- •346
183.73	- .569
Coast Periods	
First Stage Coast	7.42 sec
Second Stage Coast	22.39 sec
Third Stage Coast	3.33 sec
Fourth Stage Coast	9.5 sec
Launch Azimuth	90 deg
Launch Position:	
Geodetic Latitude	37.8479 deg
Longitude	-75.4739 deg
Altitude	0.0 feet

TABLE 11. ESCAPE MISSION INPUT DATA AND NOMINAL TRAJECTORY PITCH PROFILE: (Cont'd)

Fifth-Stage Data:

time-sec	lbs Cons Wt remaining	lbs Thrust	Sta. c.gin	Roll, I _x	Pitch (y
0.00	194.85	0.00	15.36	2.62	12.57
0.82	179.35	5905.0	14.88	2.56	12.23
1.67	162.80	6080.0	14.40	2.46	11.88
2.53	145.70	6455.0	13.92	2.36	11.54
3.42	128.23	6210.0	13.32	2.24	11.18
4.33	110.98	6030.0	12.72	2.10	10.86
5.19	93.93	5975.0	12.12	1.97	10.62
6.07	76.85	5960.0	11.40	1.80	10.36
6.95	59.94	5940.0	10.80	1.66	10.12
7.81	42.99	5920.0	9.96	1.48	9.89
8.69	26.00	5600.0	9.00	1.30	9.65
9.55	9.13	50.0	7.92	1.10	9.42
10.01	- 0.00	0.0	7.56	.98	9.42

Nominal Conditions at Fifth Stage Burnout

Time: 240 sec

Altitude: 679,627 ft

Velocity: 39,199 ft/sec

Flight Path Angle 11.77 deg

Semi-major Axis -60.673×10^8 ft

Eccentricity 1.342

Inclination 37.66 deg

Inertial Longitude 293.59 deg

Declination 37.317 deg

TABLE 11. STATISTICAL SUMMARY OF THE ESCAPE MISSION ERROR ANALYSIS RESULTS

DISPERSION OF THE SEMIMAJOR AXIS(FT.)

	MEAN	#.	-5.4098694D	04
	STANDARD DEVIAT	ION =	1.8169138D	06
	SMALLEST SAMPLE	anna an aige ann an 🗯	-5.0291464E	06
	2ND PERCENTILE	SAMPLE =	-3.9395505E	06
	5TH PERCENTILE	SAMPLE =	-2.9435135E	06
	95TH PERCENTILE	SAMPLE ,=	2.7749790E	06
	98TH PERCENTILE	SAMPLE =	3.5809730E	06
r transfer	LARGEST SAMPLE		6.0420049E	06

ECCENTRICITY DISPERSION

: 30 TO TO THE SECOND STATE OF THE SECOND STAT	MEAN		1.10235060-05
	STANDARD DEVIATION	#	1.0178485D-02
	SMALLEST SAMPLE	-	-2.6119009E-02
	2ND PERCENTILE SAMPLE	*	-2.0592809E-02
	5TH PERCENTILE SAMPLE	=	-1.5223548E-02
	95TH PERCENTILE SAMPLE	=	1.6832843E-02
	98TH PERCENTILE SAMPLE		2.0971969E-02
	LARGEST SAMPLE		3.9007366E-02

INCL	.INATION	DISPERS	NOIS	(DEGREES)		
	IEAN				1.6885141D-02	
	TANDARD	DEVIATI	ON	2	4.5528176D-02	
	MALLEST	SAMPLE		.=	-1.8545580E-01	
	2ND PERC	ENTILE	SAMPLE	=	-7.0724010E-02	
	5TH PERC	ENTILE	SAMPLE	. 7.4. N. ¥ 5.5.	-5.6196213E-02	
Committee of the second	5TH PERC	ENTILE	SAMPLE		8.7185383E-02	
	18TH PERC	ENTILE	SAMPLE		1.0744667E-01	
\$	ARGEST S	AMPLE		#	2.3364782E-01	

	LONG. OF ASCENDING NODE DI	SPERSIO	N (DEGREES)	16
	MEAN		3.6453081D-02	
	STANDARD DEVIATION	*	5.04302210-01	
	SMALLEST SAMPLE	=	-3.2027664E 00	
	2ND PERCENTILE SAMPLE	=	-9.4781303E-01	ri.
	5TH PERCENTILE SAMPLE	12 (C) - 17 (C)	-7.3487282E-01	
	95TH PERCENTILE SAMPLE		8.6254501E-01	
	98TH PERCENTILE SAMPLE	=	1.0262108E 00	
. , , , , , , , , , , , , , , ,	LARGEST SAMPLE	*	1.5043888E 00	

ARGUMENT OF P	ERIGEE DISPERS	ION	(DEGREES)
MEAN			-3.6826250D-02
STANDARD DE	VIATION	=	7.6414563D-01
SMALLEST SA	MPLE	=	-2.2351227E 00
2ND PERCEN	TILE SAMPLE	=	-1.5841312E 00
5TH PERCEN	ITILE SAMPLE	*	-1.2836466E 00
95TH PERCEN	ITILE SAMPLE	=	1.1474323E 00
98TH PERCEN	ITILE SAMPLE	=	1.4977131E 00
LARGEST SAM	IPLE	.=	2.7279234E Q0

STANDARD DEVIATION SMALLEST SAMPLE 2ND PERCENTILE SAMPLE 5TH PERCENTILE SAMPLE 95TH PERCENTILE SAMPLE 98TH PERCENTILE SAMPLE LARGEST SAMPLE RADIUS VECTOR DISPERSION (F		2.3955449D -7.0836731E -4.9756896E -4.0018737E 3.6396118E 4.4561157E 1.5092627E	01 01 01 01 01
2ND PERCENTILE SAMPLE 5TH PERCENTILE SAMPLE 95TH PERCENTILE SAMPLE 98TH PERCENTILE SAMPLE LARGEST SAMPLE RADIUS VECTOR DISPERSION (F	= = = = = = = = = = = = = = = = = = = =	-4.9756896E -4.0018737E 3.6396118E 4.4561157E	01 01 01 01
5TH PERCENTILE SAMPLE 95TH PERCENTILE SAMPLE 98TH PERCENTILE SAMPLE LARGEST SAMPLE RADIUS VECTOR DISPERSION (F	= = = = = = = = = = = = = = = = = = = =	-4.0018737E 3.6396118E 4.4561157E	01 01 01
95TH PERCENTILE SAMPLE 98TH PERCENTILE SAMPLE LARGEST SAMPLE RADIUS VECTOR DISPERSION (F	=	3.6396118E 4.4561157E	01 01
98TH PERCENTILE SAMPLE LARGEST SAMPLE RADIUS VECTOR DISPERSION (F	=	4.4561157E	01
LARGEST SAMPLE RADIUS VECTOR DISPERSION (F	=		
RADIUS VECTOR DISPERSION (F		1.5092627E	02
MEAN	() () () () () () () () () ()		
CTANDADD DEVIATION	=	2.58222190	
	=	3.05581750	07E 15
SMALLEST SAMPLE	=	-9.3401500E	
2ND PERCENTILE SAMPLE 5TH PERCENTILE SAMPLE	3	-6.2241500E -4.6287500E	
95TH PERCENTILE SAMPLE	_	5.1164250E	
98TH PERCENTILE SAMPLE		6.7577500E	
LARGEST SAMPLE	=	9.8796250E	ACCORDANCE OF THE STATE OF THE
INERTIAL VELOCITY DISPERSION	20 2 6 /8	PS)	
MEAN	_	-2.0226798D	00
STANDARD DEVIATION	=	7.8724198D	
SMALLEST SAMPLE	=	-2.3464990E	
2ND PERCENTILE SAMPLE	=	-1.5807861E	02
5TH PERCENTILE SAMPLE	-	-1.3157764E	.02
95TH PERCENTILE SAMPLE	=	1.2882373E	
98TH PERCENTILE SAMPLE	=	1.6031934E	
LARGEST SAMPLE	=	2.5061621E	02
AIRSPEED DISPERSION (FPS)			
	* 1		•
MEAN	-	-1.82980520	
STANDARD DEVIATION	=	7.89456110	
SMALLEST SAMPLE	=	-2.3756836E	
2ND PERCENTILE SAMPLE 5TH PERCENTILE SAMPLE	=	-1.5806348E -1.3037890E	
95TH PERCENTILE SAMPLE	_	1.2856494E	
98TH PERCENTILE SAMPLE	_	1.6292920E	
LARGEST SAMPLE	=	2.4808057E	
INERTIAL FLIGHT PATH ANGLE DI	coci	RSIDN (DEGRI	
	or Li		
MEAN	#	-7.2390754D	
STANDARD DEVIATION	#	3.7744043D	1 (2) ₁₁
SMALLEST SAMPLE	=	-9.6414053E	
2ND PERCENTILE SAMPLE	=	-7.5141764E	
5TH PERCENTILE SAMPLE	=	-6.7253959E	
95TH PERCENTILE SAMPLE		6.2411928E	
98TH PERCENTILE SAMPLE LARGEST SAMPLE	=	7.4625742E- 1.4544786E	Same to the control of the control o

e de Constantin e de la compania de	
ATMOSPHERIC FLIGHT PATH AND	GLE DISPERSION (DEGREES)
Advisory Commence of the Comme	
MEAN STANDARD REVEATION	= -7.4424780D-03
STANDARD DEVIATION	= 3.8957668D-01
SMALLEST SAMPLE 2ND PERCENTILE SAMPLE	= -9.9504232E-01 = -7.7478158E-01
5TH PERCENTILE SAMPLE	= -6.9403327E-01
95TH PERCENTILE SAMPLE	= 6.4494646E-01
98TH PERCENTILE SAMPLE	= 7.6998055E-01
LARGEST SAMPLE	= 1.5011804E 00
APOGEE DISPERSION (NM)	
MEAN	= -1.79393280 01
STANDARD DEVIATION	= 5.9938976D 02
SMALLEST SAMPLE	= -1.6566194E 03
2ND PERCENTILE SAMPLE	= -1.3063313E 03
5TH PERCENTILE SAMPLE	= -9.7197704E 02
95TH PERCENTILE SAMPLE	= 9.2136986E 02
98TH PERCENTILE SAMPLE	= 1.1822742E 03
LARGEST SAMPLE	= 1.9788460E 03
PERIGEE DISPERSION (NM)	
MEAN	= 1.3228584D-01
STANDARD DEVIATION	= 6.3033234D 00
SMALLEST SAMPLE	= -1.6904144E 01
2ND PERCENTILE SAMPLE	= -1.2525665E 01
5TH PERCENTILE SAMPLE	= -1.0742615E 01
95TH PERCENTILE SAMPLE	= 1.0363922E 01
98TH PERCENTILE SAMPLE	= 1.3788238E 01
LARGEST SAMPLE	= 2.2240722E 01
PERIOD DISPERSION (SEC)	
MEAN	= -4.18818470 01
STANDARD DEVIATION	= 1.12449490 03
SMALLEST SAMPLE	= -3.1755149E 03
2ND PERCENTILE SAMPLE 5TH PERCENTILE SAMPLE	= -2.4768352E 03 = -1.8432788E 03
95TH PERCENTILE SAMPLE	= 1.6972954E 03
98TH PERCENTILE SAMPLE	= 2.1827852E 03
LARGEST SAMPLE	= 3.6439482E 03
A CONTRACTOR OF THE CONTRACTOR	
LONGITUDE DISPESSION /DE	
LQNGITUDE DISPERSION (DEC	GREES)
MEAN	= 7.14427890-03
STANDARD DEVIATION	= 2.8098973D-02
SMALLEST SAMPLE	= -7.8540802E-02
2ND PERCENTILE SAMPLE	= -4.8784256E-02
5TH PERCENTILE SAMPLE	= -4.0257454E-02
95TH PERCENTILE SAMPLE 98TH PERCENTILE SAMPLE	= 5.4620743E-02
LARGEST SAMPLE	= 6.5712929E-02 = 1.0338879E-01
Eniver Chille	ATTOUCH TO THE TANK

LATITUDE DISPERSION (DGREE	SJ	
i MEAN	-	1.8003647D-02
STANDARD DEVIATION	=	4.45715650-02
SMALLEST SAMPLE	₩.	-2.1982241E-01
2ND PERCENTILE SAMPLE	=	-7.1645737E-02
5TH PERCENTILE SAMPLE	=	-5.7895660E-02
95TH PERCENTILE SAMPLE	=	8.9109421E-02
98TH PERCENTILE SAMPLE	=	1.0219049E-01
LARGEST SAMPLE	=	1.5267134E-01

ALTITUDE DISPERSION (NM)		
MEAN	-	4-24979510-01
STANDARD DEVIATION	=	5.0292293D 00
SMALLEST SAMPLE	=	-1.5371993E 01
2ND PERCENTILE SAMPLE	=	-1.0243593E 01
5TH PERCENTILE SAMPLE	. +	-7.6180258E 00
95TH PERCENTILE SAMPLE	. I i = 1	8.4205933E 00
98TH PERCENTILE SAMPLE	=	1.1121825E 01
LARGEST SAMPLE	=	1.6259771E 01

TABLE 13. SIGNIFICANT ERROR SOURCES AND THEIR THREE SIGMA CONTRIBUTIONS AT FIFTH STAGE BURNOUT

Error Source	lo Mag.	Altitude (feet)	Velocity (ft/sec)	Flt. Path Angle (deg)	Inclination (deg)
SIW1 First Stage	.83% Inert Weight	8720		.0557	
ISP1 Specific Im	.18% pulse - First S	15973 tage	18.07	.1059	
MFR1 Flow Rate -	1.4% First Stage	18304	19.71	.1056	
DKSG Torquer Sca	.35% le Factor	6156	24.36	.0826	
TMP1 Pitch Thrus	1.67 mrad t Misalignment	14741 - First Stage	-65.43		.0639
TMYl Yaw Thrust	1.67 mrad Misalignment -				
CAO1 Drag Coeffi	1% cient	6615			
DRHO Density	6.67%	4218		.2673	
FWN1 Wind Effect	s				.0179
TIM1 Timer Step	.078 sec Uncertainty	12343	28.96	.0739	
ISP2 Specific Im	.094% pulse - Second	6401 Stage			
MFR2 Flow Rate -	1% Second Stage	7002	16.63	.0596	
DBY2 Yaw Dead Ba	10% nd - Second Sta	ıge			.0136
TMP2 Pitch Thrus	1.67 mrad t Misalignment	-51023 - Second Stag	54 . 89 e	0349	
TMY2 Yaw Thrust	1.67 mrad Misalignment -	Second Stage			.0554
ISP3 Specific Im	.14% pulse - Third S	tage	27.17	.0687	

TABLE 13. (CONTINUED)

Error Source	O	Altitude (feet)		Flt. Path Angle (deg)	Inclination (deg)
MFR3 Flow Rate	1.8% - Third Stage		24.44		
TMP3 Pitch Thru	.557 mrad st Misalignment	14860 - Third Stag	41 . 95	.1194	
TMY3 Yaw Thrust	.557 mrad Misalignment -	Third Stage			0136
ISP4 Specific I	.6% mpulse - Fourth	Stage	14.99		
	.03 rad/sec ng Rate - Fourth	Stage	17.41	0633	.0467
W4CY Yaw Coning	.03 rad/sec Rate - Fourth S	tage	69.88	.5886	
MFR5 Flow Rate	1.8% - Fifth Stage		-52.66	1054	
ISP 5 Specific I	.15% mpulse - Fifth S	tage	48,94		
W5CP Pitch Coni	.0463 rad/sec	Stage	-19.77		.0719
W5CY Yaw Coning	.0463 rad/sec Rate - Fifth St	age	-25.71	.5726	

TABLE 14. INDIVIDUAL THREE SIGMA ERROR SOURCE CONTRIBUTIONS AT SECOND STAGE IGNITION

INCL.DEV.	O	.293397E-	5.549529E-06		07	1.054410E-04	.182	• •	-8.537736E-07	• 0	.494104E-0	11	-158962E-	-2.988208E-06	-3.841981E-06	0849E-0	708E-	4.490849E-04	1	1	7	.537736E-0		.537736E-		1.011722E-04	-8.537736E-07	736E-0	247E-0	-1.707547E-06	.707547E-	7.345868E-03	700005	0-36767	4530445-0	0-3000660	-4/000/E-	• 792925E-	-537736E-	-8.531/36E-01	1,0000	-8-53//36E-0/	
FLT. PATH ANGLE DEV. (DEG)	20854E	.63921	8.704457E-02	.165814E-0	6.471177E-03	.307242E-0	3.5789496-01	-9.969941E-04	9.063447E-03	98166E-0	-1.269775E-03	위	9.961404E-0	-7.282689E-04	4.839189E-03	-1.749382E-03	.201494E-0	-1.838431E-02	.726804E-0	2.508605E-01	-2.655236E-04	•	2.810623E-03		-121627E-0	186E	4.268868E-07	•	.028797E-0	9	.070401E-0	1.385622E 00	0 330505	9714345-0	0-10011	-27.20.27.	0-300C+69*	-544175E-0	0		•522866E-0	18208	
VELOCITY DEV. (FPS)	92822E-	1.571960E	-5.292358E-01	.937866E-0	-3.967285E-02	.703149E 0	36	.029	.60827	.1990)E-0	8	.133179E-0	16	.549316E-0	.361	.508142E	.795410E 0	.415039E 0	-5.069885E 00	5.212402E-02	•0	5.249023E-01	•	五	-6.798096E-01	.051758E-0	•0	-165771E-0	2793E-0	.455566E-0	-4-425409E 01	0 701100	3030000	0.30233010	102/16-0	10-3888860°/	.619141E-0	-0-		-553223E-	9912E-0	
RANGE DEVIATION	0	.003750E 0		Ç	0	.940500E 0	2.372000E 03	.500000E-0	.475000E 0	0.	•	3.375000E 01		•0	2.450000E 01	.0	.180000E 0	E 0	1.00000CE CO	O	6.	o	3.875000E 01	0.	0.				5.00000CE-01	ô	9	4.379750E 03) (SSOUNCE U	Sonooc n	.000000E 0			20000E	•	
STND.DEV.		8.300000E-03	4.100000E-03	9.30C000E-C4	2.400000E-03	800000E-0	1.400000E-02	1.03000E-07	1.080000E-07	1.C30000E-C7	9.250000E-05	5.760000E-05	.430000E	3.280000E-06	.250000E	. C00000E	3.500000E-03	3.570000E-03	3.570000E-03	3.57G00CE-03	1.450000E-03	1.450000E-03	2.430000E-02	3.060000E-02	2.330000E-02	4-400000E-02	2.100000E-02		.130000E-1	3.120000E-11	.130000E	1.67600E-03		5019000000		. CCOCCOE-C	200000	-300000E	COOOCE	• 000000e	GOOCOGE	Z.C00000E-03	
CODE			3 SIN2		S		7 MFR1	8 KRIA		10 KYIA	11 THOR	12 THOP	13 TH0Y			1	17 DKSG		19 DYBE	1	21 TYRG	22 TRRG		24 KPRI		W 8 2	27 KRY1	28 KRR1				32 TMP1	Xii	24 CAUL		- 15	20.33	10.77	- 4		2	42 CRUI	73

	¥		-03 -1.327618E-04	.067217E-	-06 -8.537736E-07	00 1.861227E-03	-298359E-	4 -2.988208E-	-05 -5.976415E-06	-2.988208E-	-2.134434E-	-06 -8.537736E-07	-8.537736E-0	.695755E-0	-03 8.153538E-05	-2.988208E-0	.086321	-04 -3.756604E-05	38.3	2.0	335.di	49529E-0	.677595	0	91	-9.391510E-0	-04-695755E-06
The state of the s	7.918750E-	1-562775E-	-5.581118E-	.157077	9	1.276965E	-3165160°5	1.227300E-	-1.878302E-	5283E	.077889E	1	-2.774764E-		6.979813E-	1.696534E-	1.774946E-	-3.590118E-	-2.327771E-	-9.554175E-	-4.521372E-	.521372	2.038619E-	8.138922E-	1.839477E-	8.083102E-	4-014871E-
	-481934E-0	-3.123047E 00	7.241211E-01	3.114014E-01	1.525879E-03	-1.749060E 01	322	1.287842E-02		2.624512E-03	2.160645E-02	1.831055E-04	5.493164E-04	1.146851E-01	-4.747314E-01	-2.426147E-01	-2.722107E 00	5.401611E-02	.345520E	-7.635254E 00	-3.633179E 00	-3.633179E 00	3.108521E-01	53	2.082764E 00	.29	4-595947E-02
	4.000000E 00		-6.750000E 00	1.250000E 00	1	ш	3.150000E 01	7.50000E-01	٥	-2.500000E-01	•0	•0	•0	5.000000E-01	9.750000E.00	E O	5.035000E 02	-5.00000CE-01	-2.750000E 00	-5.635000E 02	-2.657500E 02	-2.657500E 02	6.77500CE 01	2.687500E 03	0	0	1.500000E 00
	2.000000E-03	9.000000E-02	3.30000E-01	3.30000E-01	3.300000E-01	6.67000E-02	10.000000E-01	5.760000E-05	5-760000E-05	5.760000E-05	5.760000E-C5	•	5.760000E-05	10.00000E-03	10.000000E-C3	1.C00000E-01	1.000000E-01	1.000000E-01	1.000000E-01	1.0000006-01	1.000000E-01	1.C00000E-01	1.000000E-01	7.800000E-02	4.000000E-03	3.000000E-03	3.000000E-03
4	43 CHAL	44 CM01	45 NCBA	1	47 NCR1		49 FWN1	50 CNDR	27.00	52 CLDR	53 CLD0		55 NC00	56 LSMY	57 LSMP	1	59 MSMR		61 NSMR		63 CDV2	64 CDV3	65 LDA2	66 TIMI	67 TIM2	-	PALL 69

TABLE 15. INDIVIDUAL THREE SIGMA ERROR SOURCE CONTRIBUTIONS AT THIRD STAGE IGNITION

-3.841981E-06 1.169670E-04	.351510E-0	٠		-227069E-0	.134434E-	-8-33/138E-06	-200000*	-2.134434E-06	0660E-0	20896E-	-	.537736E-0	25434E-0	.257076E-0	.380979E-0	1986 1986	287028E-0	2099 1E-	.670071E-0	.201651E-0	.549529E-	0-41581F-C	.478043E-	-1.0(3)(3)E-04 -8.044433E-05	9.090128E-0	-36C579E-	-4-764057E-04	-113481E-	-180342E-	04835E-	3726E-	42335E-	-2.35 110F=04	.741628E-	.203750E-	-1.622170E-04	
3.671227E-04 5.512646E-02 -4.235784E-04	.124706E-0	.604953E-0	-814220E-0	8	-029116E-0	0673175-	00170700	8.537736E-07	•	.125826	.950873E-0	.092422E-0	.340721E-0	04069E-0	.201793E-0	52556E	157518E-0	-185448E-0	862278E-0	-101761E-0	.983921E-0	9-4610898	-984989E-	787440E-0	.268883E-0	-356337E-	47111E-0	.942282E-0	-245354E-	37514E 0	.274649E-	23986E-0	-7-796341F-04	036998E-0	.804931E-	-1.243308E-04	888873
5.261230E-02 -4.672363E 00 -2.801514E-01	.020508E-	82813E-0	17E 0	06372	219E-0	1035165-0	0-10776010	1.220703E-04		-248047	.277832E-0	.090576E-0	.470093E 0	6914E-0	.175659E 0	89258E 0	0881335	*344727E-0	.343286E 0	.770020E 0	22363	0-4/64816.	.586914E 0	0 301020	.278296E 0	-270874E 0	52539E-0	.269409E 0	1299E-	.720410E 0	59094	.512451E-0	-2.139282F 00	-241455E-0	16064E-	.263428E-	
9.250000E 00 1.05275CE 03		.50000CE-0	538250E C	5000E 0	suuduue o	-3000005	•		•0	-000000E		OCCE C	24750E C	9	.75000CE 0	.19500CE 0	יו שו	.67750CE C	.90825CE 0	0 3	(4)	COCCE	.007500E C	D SUCCOSE O	.47500CE C	.38250CE C	0.	00000E 0	12	-72100CE 0	.110000E C	•	200000F C		.75000CE 0	•0	
2.000000E-03 9.000000E-02	-30000E-	000000	.670000E-0	-000000-		740000E-C	0-10000F		5.760000E-05	-0000000-	10.000000E+03	-0000000-	-30000	1		1.CGGGGE-01		00000E-0	800000E	-000000E	100	4000000.	400000 400000		-400000E-0	-Coocook	2.620000E-02	-3000000.	COCCOE-C	1.67000E-C3	. 6 70000E-C	3.300000E-03 v	000	C0000E-	00000E-0	3.000000E-03	
43 CHAL 44 CHOI 45 NGBA	46 NCDR		-	49 FWN1		51 C100			1000	56 LSHY	57 LSMP			- 1	24 9. 13.	62 CDVI	334				68 TIM3	541- 50		72 MED2	KRP		30,063		77 RDE2	- 1	3 72	80 C2PY	2.0	CNA	7 4	85 TIM6 86 TIM7	

TABLE 16. INDIVIDUAL THREE SIGMA ERROR SOURCE CONTRIBUTIONS AT FOURTH STAGE IGNITION

V. INCL.DEV.	3.713915E-05 2.228349E-04	-299765E-0	*646698E-0	-695755E-	.801830	-38113E-	716	.549529	-918821E-	28	.9646	-3.722453E-04	•	-7-257076E-06	-980684E-	.941250E-	.9220	448	.117359E-	8821E-0	.537736E-0	-9.818397E-06	\$ pr	.287028E-	0	•0		37736E-	•134434E-	9	.553868E-0	•623006E-0	-892052E-	.246510E-0	.536793E-	.336085E-	• 15				
FLT. PATH ANGLE DEV	926 822	(1)	.298728E-	52675E-0	.963 (54t-0	49 IE-0	.865390E-0	•032942E-0	09 61 1E-0	*844045E-0	.100390E-0	.114175E-0	895342E-0	•784280E-0	.754364E-0	93740E-0	0196E-0	.836360E-	.454984E-0	1.803597E-05	.067217E-0	.070205E-0		.355366E-0		.403302E-0	•	3.756604E-05	•	302E-0	.356105E-	*596604E-0	50 90 0E-0	-606770E-	613E-0	-138650E-	.967807E-0	0	1.4311385-04	1	
VELOCITY DEV.	992E-	,	.363770E-	.098633E-0	.946997E	2.072827E 01	.709961E-	•861328E-	785	.491211E-	-418457E-	.344727E-	.879883E-	.297852E-	1.	.394214€ 0	.227051E-	109E-0	*320068E	7.324219E-03	-441406E-	.409180		.371094E-	57129E	٠,		.098633E-	41406E-	.882813E-	-6.120654E 01	.383057E	1553E 0	.968506E 0	38	1621	0	0	4. 467773E-02	0-101101.	
RANGE DEVIATION	.164	66250E	15000E	500	.146200E	.248250E				6.50000E 00	1.170000E 02	8.50000E 00	.250000E	.000000E	.500000E		9.825000E 01	300	.873250E	1.250000E 00		5.400000E 01	•0	7.500000E-01	662500E	.0	•	2.000000E 00	•0		1.176375E 04	.985750E	.746750E	.492750E		.1500		652.23		0 3000000	
STND. DEV.	6. 000000E-04 8.300000E-03	.100000	.300000E-0	E-0	11		1	.080000E-	1.030000E-07	9.250000E-05	-760000E-	4	280000E-	.250000E-	-3000000 ·	3.500000E-03	.570000E-	.570000E-		.450000E	.450000E	430000E-	060000E-	1	400000E-0	1	.620000E-0	.130000E-1	.120000E-	*130000E-1	.670000E-	.67000E-0	.000000E-0	00000E-0	2.000000E-01	-300000E-	山	-000000-	1.00000E-03	• 000000	
CODE	1 PWIO 2 SIWI	3 SINZ	4 SIM3	5 SIM4			8 KRIA	9 KPSA	10 KYIA		12 TH0P	13 TH0Y	14 DTER	74.		17 DKSG	18 DRBE			21 TYRG	22 TRRG		24 KPR1		26 KRP1	27 KRY1				31 KYAN	1	X 9.3			36 CNDQ	12.5	38 CYDR		֓֞֝֝֓֞֜֜֝֓֓֓֓֓֓֓֓֓֓֓֓֓֓֓֓֡֓֓֓֓֡֓֡֓֓֓֡֓֡֓֡֓֡֓֡֡֓֡	5	79

-0 5377365-07	707547E-	· 中	8355	-185661E-	37736E	81724E	0	-8.537736E-07	*	2642E-0	36793E-	-8.537736E-07		.716	-2.830260E-04	-39066	23E-	•6046	-372833E-	.0885	330E-	-165330E-	1	1 1	.518632E-	1		.595401E-	.527170E-		.742712E-	19594E-	.423562E-0	1	-301774E-	-647783E-	-3.233155E-02	2.881486E-04	3	•415095E-	96245E-	9.895236E-04
4 30 3	0393675	76899E-0	.156722E-0	-	2.134434E-06	.2	8.380108E-03	4.642394E-05	4.589033E-06	.53773	.3637	2.134434E-07	34434E	.420466E-0	2.096014E-03	.025436E-0	4.131891E-02	.400118	*614558E-0	-5.627670E-02		*669878E-0	2E-0	1.794155E-01	6.851320E-03	2-972199E-04	1.465289E-04	*525768E-	.202737E-0	.758067E-0	*383309E-0	.717342E-0	.091710E-0	•615091E-0	-428168E-	10	-3.030459E-02	1.484499E-04	-665926E-0	.216051E-	.05103	-2.767294E-03
7 7 7 7 7 7 7 7 7 7 7 7 7 7 7 7 7 7 7 7	7307915	0401C	-628418E-	4063	4	345020E 0	7	***	7.324219E-04	2.441406E-04	2.441406E-03	.441406E-0	•0	2.612305E-02	1.513672E-02	-4.831543E-01	-5.218262E 00	3.417969E-03	6445E-	.520020E 0	-2.629883E 00	-2.629883E 00	-1.218262E-01	-2.634839E 01	1.989014E 00	8.886719E-02	*418945E-0	0	.497461E 0	0	Ш	.982666E 0	.782227E-	.018555E 0	.014160E-	4.661865E 01	-8.411621E 00	7.055664E-02	-812500E-	.029053E	-821777E-	3.449707E-01
(2.50000E 00	1.771000F 03	0	65000E 0	• 0		.505000E	2.500000E 00	.500000E-	0.	5.000000E-01	•0	•0	9.00000E 00	1.182500E 02	147500E 0	2.192500E 03	-2.750000E 00	6.425000E 01	-3.277000E 03	-1.554500E 03	-1.554500E 03	3.887500E 02	9.469250E 03	3.842500E 02	1.650000E 01	.250000E	-8.307500E 02	LLI.	ü	111	-737000E	ш		-3.312500E 02	-3.764900E 04	-1.714500E 03	8.750000E 00	DODOOE	.632500E	22500E	-1.305000E 02
	2.00000E=03	2. UduquuE=03 9. ODOQQQE=02	300000E-	-300000E-	-300000E-	.6 70000E-	-0000000.	5. 760000E-05	5. 760000E-05	-760000E-	5. 760000E-05	-	5. 760000E-05	10.000000E-03	10.00000E-03	1.000000E-01	1.000000E-01	0000000	1.000000E-01	1.000000E-01	1.000000E-01	1.000000E-01	1.000000E-01		4.000000E-03	3.000000E-03	-000000.	5.400000E-04	9.400000E-04	10.000000E-03	4.400000E-02	000000E-	2.620000E-02	1.000000E-01	2.500000E-01	1.670000E-03	1.670000E-03	3,3000006-03	300000E	1	1	1-000000E-01
80		43 CMAL		3				50 CNDR	100		53 CL00		55 NCDQ		57 LSMP					S/A	63 C0V2	64 CDV3			67 TIM2	68 TIM3	y.,	70 PW20	71 ISP2	72 MFR2	73 KRP2		75 KRY2	76 0BY2	77 ROE2	78 TMP2		.80 C2PY	81 C2VP		5	84 ZET2

2.399104E-04	-3.675495E-04	• 0	-6.650897E-04
0.	-3.171769E-04		-1.246510E-04
5.805661E-05	-8.537736E-07		-7.742873E-03
1.272123E-04	1.160437E-01	.0	-5.027659E-04
1.067217E-07	9.493621E-02		2.165460E-01
1.964864E-02	5.336085E-07		8.938583E-03
5.932617E-02	2,742993E 01	• • • • • • • • • • • • • • • • • • •	-2.319336E-01
2.441406E-04	2,270728E 01		4.096069E 01
-1.503906E 00	0,		-1.504150E 00
7.500000E 00 0. 9.375000E 02	5.928750E 03 7.541500E 03		-2.750000E 01 1.068200E 04 4.172500E 02
3.000000E-03 3.000000E-03 6.000000E-04	1.400000E-03 1.800000E-02	1.000000E-01 2.100000E-02 1.000000E-01	2.500000E-01 5.570000E-04 5.570000E-04
85 TIM6 86 TIM7 87 Puno	88 ISP3 89 MFR3	91 DBP3 92 KRY3 93 DBY3	

TABLE 17. INDIVIDUAL THREE SIGMA ERROR SOURCE CONTRIBUTIONS AT FIFTH STAGE IGNITION

CODE	stno.bev.	RANGE DEVIATION	0	H	
		(FT)	0042200	(UE6)	10561
	-000000-	2 (4227E 0	7183595-0	360222
	-300000E-	7666.	43080	45505F-) L
<i>^</i>	-1000001			0 1/0/1/1/1	1/0/0/07
4 SIM3	9	SOCE SOCE	0-10000000	0-305/TO7*	.007301E10.
S	2.400000E-03	.042500E 0	0-3797817	0-3706168*	.092604
6 ISP1	9	.522900E 0	.836206E 0	.436122E-0	.102382E-
7 MFR1	1.400000E-02	55975E 0	.997583E 0	.430686E-0	.702194E-
8 KRIA	1.030000E-07	.500000E 0	.757813E-0	.328597E-0	.920991E-
		.35000E 0	.134766E-0	-280447E-0	-134434E-
	030000E-	000000 O	.026367E-0	.845165E-0	.815283E-
100	250000E-	.000000E 0	.368164E-0	.445991E-0	.3873
1	5.76000E-05	E 0	.567383E-0	.599758E-0	.403302E-
	-430000E-	.150000E 0	.416992E-0	.708491E-0	.571958E-0
4 DTER	.280000E-0	.50000E 0	.269531E-0	.287028E-0	.077854E-
1	-250000E-	.257500E 0	.424805E-0	.353445E-0	.695755
7-1	000000	150000E 0	.173828E-0	.142866E-0	.338184E-
7 DKSG	. 500000E	.792000E 0	.451294E 0	.212836E-0	.677665E-0
	.570000E-0	0	.503418E-0	.256968E-0	.650736E-0
	570000E-0	42500E 0	*997559E-0	.514701E-0	.409510E-
		.597500E 0	.070068E 0	.396173E-0	.229434E-0
TYR	.450000E-0	00000E 0	.126953E-0	.440743E-0	-152594E-
	4			.134434E-0	.134434E-0
	430000E-0	6.425000E 01	4.470215E-01	.533467E-0	.830189E-
	-060000E-		.0	•0	.134434E-0
	-33000E-0		0-3696115°	.120578E-0	.012736E-
¥	-400000E-	4.575000E 02		.300351E-	.536793E-0
7 KRYI	2.100000E-02	·o	•0	.067217E-0	-134634E-
8 KRR1	2.620000E-02	•	•0	•	.134434E-
315	.130000E-	2.50000E 00	1.147461E-02	.8814	.134434E-
O KRAN	7.	•0	.441406E-0	.201651E-0	.707547E-0
1 KYAN	3.130000E-11	•0	.441406E-0	.336085E-0	.561321
2 TMP1	9		.530664E 0	.352102E-0	.00105
		0	.367432E 0	*152229E-0	-941052
4 CA01	10.00000E-03	.322500E 0	.457520E 0	.876823E-0	.7673
Ü	-000000E-	.159500E 0	.535889E 0	-962616E-0	.232548
	-0000000.	.872500	6.250000E-01	16692E-	-8.964623E-06
7 CYBA	3.300000E-01	000E 0	.642578E-0	.964711E-0	-672241E-
8 CYDR	3.300000E-01	50000E 0	*765625E-0	.064977E-0	-2.160047E-04
T) 6	-3000C	•0	•0	•	-2.134434E-06
	10000000	C	C	C	

CMC	2, 400000E-03	1.25000CE 01	844E-0	.124706E-0	-537736E-
LON J	00000E-0	0	42773E-0	.283844E-0	.537736E-0
Z W Z I	-000000F-	.25000E 0	-933594E-	.261433E-0	.134434E-0
J CW J	.000000E-0	.218250E 0	5869E 0	602E-	0519E-0
NCRA	-300000E-	4	.271973E-0	.920902E-0	-966ED09.
NCOR	4	225000E 0	96875E-0	.103485E-0	.744882E-0
NCD1	-300000E-			.494104	
OBHO	-67000F-	.034550E 0	53223E 0	.724113E-0	45757E-
	COOCOE	. 8250	8535E 0	1E-	.441255E-
ACN C	760000F-	.00000E	269531E-0	.553833E-0	37736E-
CYDO	760000E-	500000E-0	24219E-0	.094929E-0	-30990
9000	76000F		1406E-0	.3360	07547E-0
CLUA	76000E-	5-0000005-01	64844E-	76415E-0	-963679E-
ONU	- 760000F-		•0	-336085E-0	34434E-
	-760000F		•0	.067217E-	.537736E-
NW N	-100000E-	1.20000E 01	2.124023E-02	.053343E-0	.108821E-
CMD	-300000	000F 0	6875E-0	-484072E-0	-590118E-
Long	3000000	717500F O	-346680E-	.573167E-0	-549529E-
A SELE	- 1	768750F 0	56592E	9	.086321E-
NIC MY	000000	OCF	-765625E-	.863326E-0	186745E-
NOND	00000E-	025000E 0	8906E-	.339234E-0	-217215E-
COVI	000000E-	.354500E 0	.212891	9	.165401E-
CDV2		66000E 0	83643E 0	.952676E-0	.464151E-
CDV3		0	848	52676E-	464151E-0
LDA2	1.000000E-01	EO	18945E-0	.734281E-0	.555897E
I I M I	7.800000E-02	0 3	.881226E 0	.121484E-	.880401E-0
LIM2	0000000	4.835000E 02	.036621E 0	.194572E-0	.536793E-0
TIM3	.000000E-0	ш	82031E-	2E-0	88208E-
TIME	3.000000E-03	1.025000E 01	02E-0	13107E-0	-988208E-
PW20	5.400000E-04	0	.523682E 0	-151356E-	.298680E-0
TSP2	9.400000E-04	6.061500E 03	45E 0	35824E-0	.157878E-
MFR2	10.000000E-03	0 3	93E 0	17192E-	.579481E-U
KRP2	4.400000E-02	1.580000E 02	48E	.613952E-0	·05/524E-
0802		0	.822510E 0	.504272E-0	.199517E-
KRY2		50000E 0	-39EE690°	.370271E-0	8
DRYZ	.000000	О Ш	-239502E	.089593	-095050E-
ROEZ	5000000	.357500E	09180E-	73027	.952349E-
C2PY		0 0000C	-272461	411736-	.692571E-
	-300000E-0	1.35000E 01	0742E-	89517	-046887E-
CD02		300	1729E	.184043	9099
2	-30000	3000	4-4042976-01-	-3-326089E-03	E0-326400+*I
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3.056510E-04	-2.134434E-06	7.167430E-04	-4.371108E-02	5.378774E-05	-3.513279E-04	-4.550613E-04	-2.134434E-06	-2.134434E-06	-2.134434E-06	-2.134434E-06	-8.729835E-04	-8.537736E-07	-1.033834E-02	-1.186745E-04	1.562406E-03	2.557052E-04	-1.306274E-03	1.519290E-03	6.828695E-02	6.863059E-03
1.060814E-04	3.201651E-07	-5.010121E-01	-2.356405E-02	c. 1.369175E-02	8.991965E-02	6.177714E-02	•0	•0	•0	•0	-4.205902E-04	1.577611E-01	6.008112E-03	1.898259E-03	-4.123364E-02	1.6314016-02	1.954160E-02	1.701752E-02	-8.635611E-02	8.055149E-01
3.662109E-02	•0	5.449609E 01	-5.406582E 00	-1.466309E 00	2.736597E 01	2.457471E.01	•0	•0	·O	•0	-1.630859E-01	4.230102E 01	-6.955566E-01	-3.624023E 00	1.497720E 02	9.834473E 00	∠ 2.060059E 00	1.448975E 00	-1.728101E 01	7.255347E 01
1.02500CE 01	0.	-4.872325E 04	-2.346750E 03	1.291500E 03	8.177500E 03	7.296000E 03	0.	•0	0.	• • • • • • • • • • • • • • • • • • • •	-3.85000CE 01	1,398925E 04	5.757500E 02	1.250000E 00	5.27500E 01	6.600000E 01	6.900000E 01	6.45000CE 01	-3.285000E 02	-2.851750E 03
" " " " " " " " " " " " " " " " " " "	3.00000E-03	1.670000E-03	1.670000E-03	6.000000E-04	1.400000E-03	1.800000E-02	4.400000E-02	1.000000E-01	2.100000E-02	1.000000E-01	2.500000E-01	5.570000E-04	5.570000E-04	3.400000E-04	6.000000E-03	1.800000E-02	5.00000E-04	5. C00000E-04	3.000000E-02	3.000000E-02
RA TIME	84 TIM7	85 TMP2		87 PW30	88 ISP3	89 MFR3	90 KRP3	91 08P3	92 KRY3	93 DBY3	94 R0E3	95 TMP3	96 TMY3	97. PW40	98 I SP4	99 MFR4	100 TMP4	101 TMY4	102 W4CP	103 M4CY

TABLE 18. INDIVIDUAL THREE SIGMA ERROR SOURCE CONTRIBUTIONS AT FIFTH STAGE BURNOUT

INCL DEV.	4-738444E-05 2-834528E-04	.306133E	3.372406E-05	5.549529E-06	-4.892123E-04	-3.602925E-04		4-268868E-06	-4-414010E-04	-4.947618E-04	-6.403302E-06		-2.817453E-04	-5.122642E-06	-6.740543E-04	-912453E-	4.317960E-03	6.354637E-03	1.404458E-04	-1-045873E-04	0.	-6.830189E-06		4.653066E-05		0-	•0	.268868E	3.415095E-06	-695755E-		3047766	30165E	-	-8.537736E-06	7.863255E-04		•0	• 0	•
FLI. PATH ANGLE DEV.	1.032789E-02 5.577223E-02	-839285E-	-692091E-	1.366892E-03	1.059411E-01			*499316E-0	.351061E	-503656E-0	1.269135E-03	-189859E-0	2.262500E-05	1.068711E-03	5.698939E-05	.902	9-643373E-04	7	2,231316E-02	1.0031845-05	•0	7-095926E-04		0156-0	2.813184E-03	0.	•0	47	-067217E-	5-336085E-07	-3100107		4-239509E-02		-442326E-	-997813E-	-5.261380E-05	-0-		6-211203E-05
VELOCITY DEV.	1.059570E 00 4.848633E 00		5.795898E-01	1.186523E-01	1.807.666E 01	1.970557E (0)		-3414E-	-953125E-	2.343750E-02	.528320E-	100	1.220703E-02	5.390625E-01	.125000E-	2-436328E 01	.437500E-	1	-9.109375E 00	5.371094E-03	.0	4-448242E-01	•	Pro	-155273E			1.123047E-02	•0		To second	3068367°	3.354980E 00	10141C.	-293945E-	64.	9-765625E-04	•0		4-589844E-02
RANGE DEVIATION	1.616000E 03 8.720250E(03)	2.875500E 03	1-046250E 03	2-140000E 02	1.597300E (04)	1.830400E 00	7-250000E 00			9.500000E 00			5.000000E 00	1.340000E 02		6-156500E @3	- 1		3.738250E 03	2.250000E 00	•0	6-975000E 01	1	1.250000E 00	.755000E	•0		3.000000E 00	•0		10000000 V	*077700E	6.615750E (D3	30C1007.			-7.500000E 00	••	700000	1.300000E OI
STND.DEV.	6.000000E-04	100000E	9.300000E-04	2.400000E-03	1.800000E-03	1.400000E-02		1.080000E-07	1.030000E-07	9.250000E-05	5.760000E-05	ė	3.280000E-06	1.250000E-06	1	00000	3.5 70000E-03			1.450000E-03	1.450000E-03	2.430000E-02	3. 000000E-02	2.330000E-02	*400000E	2.100000E-02	2.620000E-02	3.130000E-11	3.120000E-11	3-130000E-11	700005	3000010			2. 000000E-01		3-300000E-01	I. GOUUUUE-UI		Z-200000E-02
CODE	- 1 PALO 2 STAL	3 SIWZ	4 SIN3	5 SIM4	6 ISP1	7 MFRI		S KPSA	10 KYIA	2	12 THOP	13 THOY	14 DTER	15 DTEP	16 DTEY		- 1			21 TYRG	2	m ·	. 1	× 4		2				31 KYAN			34 CA01		36 CNDQ		38 CYUR	1	40 CLP1	¥

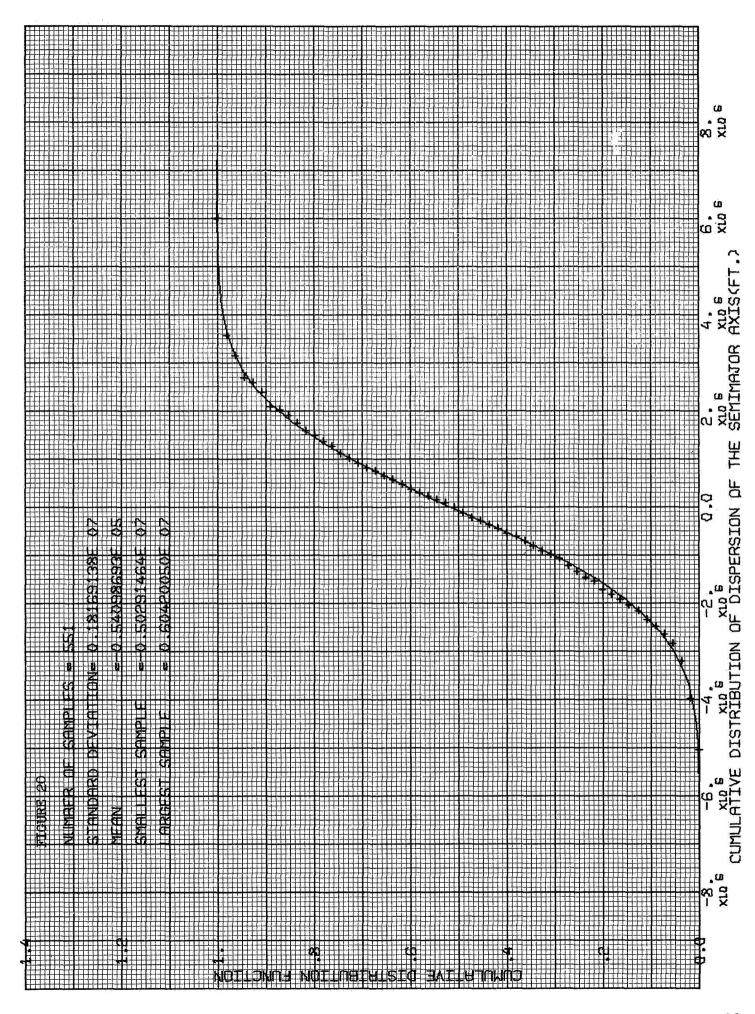
0.	-1-707547E-06	3	532I	84E-	PAF-	1.	14E-	68 E-	-024528E-0	1321E-	255E-0	0.	•0	-1-933797E-04	-4-324363E-04	7.257076E-06	004.2025	-3667490°	14555	1.3873825-04	- 702123	ة س	1934	4.418279E-04		101547	.268868	-048007	~ I	1.404458E-04	*D-13*17T*0**	4.02833312103		067100	*090803	-682948	-114104	5-370236E-04	1.762616E-03	1.589300E-03	3-884670E-04	•	9.361628F-04
1.494104E-05	1-742765E-04	796E-	-180272	81 61 6E-	-537 736E-	°673175E€	731E-0	-764092E-	-134434E-0	085E-	.589033E-0	3.201651E-07	788 888	7.523880E-05	1.062628E-03	1.727718E-03	0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0	*8 /U814E-U	-588603E-	-034,557	.440188E	-440 188E-	348393E-0	0	-097900E-	-779051E-		≈668470E	.710888E	الله	I. LES LICETUS	3 5 0000 10 000 3 5 0000 10 00	*20000-	-412331E	-187248E-	-583609E-	-072553E-	-6.915993E-03	-2-305509E-03	-1.239893E-03	8-174883E-05	1.067217E-07	-3-487 78 6E-01
1-318359E-02	938E-	-790527E	5	4.199219E-02			1.208984E 00	1		.0	1-464844E-03	0.	•0	2.099609E-02	8	-5.380859E-01	1000000 1000000	273r-	-310547E-	-5-139160E 00	•448242E	8242E	-1.992188E-01	89	-024414E	33203E-	ů	\$501953E	7861E	u		5.912109E 00			* 928984E	50391E-	867	ш	4-428711E-01	2.880859E-01	3-613281E-02	•0	5.48969ZE (01)
3.500000E 00		2.305500E 03	-7.775000E 01	2.350000E 01	•••	.4.218700E 64	7.275000E 01	3,500000E 00	5.000000E-01	0.	7.500000E-01	•0	0.	1.275000E 01		2.830000E 02	30000000	*000000E			-2.167000E 03		5.272500E 02	LLI			ш	50E	250E	ŭ	t	-6-490500E 03		-		اس		-9.482500E 02		-1.757500E 02	1.075000E 01	•0	-5.102350E (04)
2.000000F-03	-000000-		3-300000E-01	3.300000E-01	3.300000E-01	6.670000E-02	3000000	5. 760000E-05	5-760000E-05	5. 760000E-05	5. 760000E-05	5- 760000E-05	5-760000E-05	10-000000E-03		000000	10.000000°	1-000000E-01	1.000000E-01		1-000000E-01	1.000000E-01	1.000000E-01	7. 800000E-02	-3000000°	3.000000E-03	3000000	5.400000E-04	-400000 E-0		A CONTRACTOR	3000000-	<		- 2000000E-0	3.300000E-03	3.300000E-03	1.000000E-01	1. 000000E-01	1.000000E-01		3.000000E-03	L.670000E-03
S CNO		1007	45 NCBA		47 NCRI	48 DRH0	49 FUNI	SO CNDR	51 CY00	52 CLOR	4		144	LSMY		1			61 NSMR	N	63 CDV2	64 CDV3	65 LDA2		67 IIM2	***		70 PW20	emd			J. (_			4		81 CNA2	82 ZET2	149	84 TIM7	85 TMP2

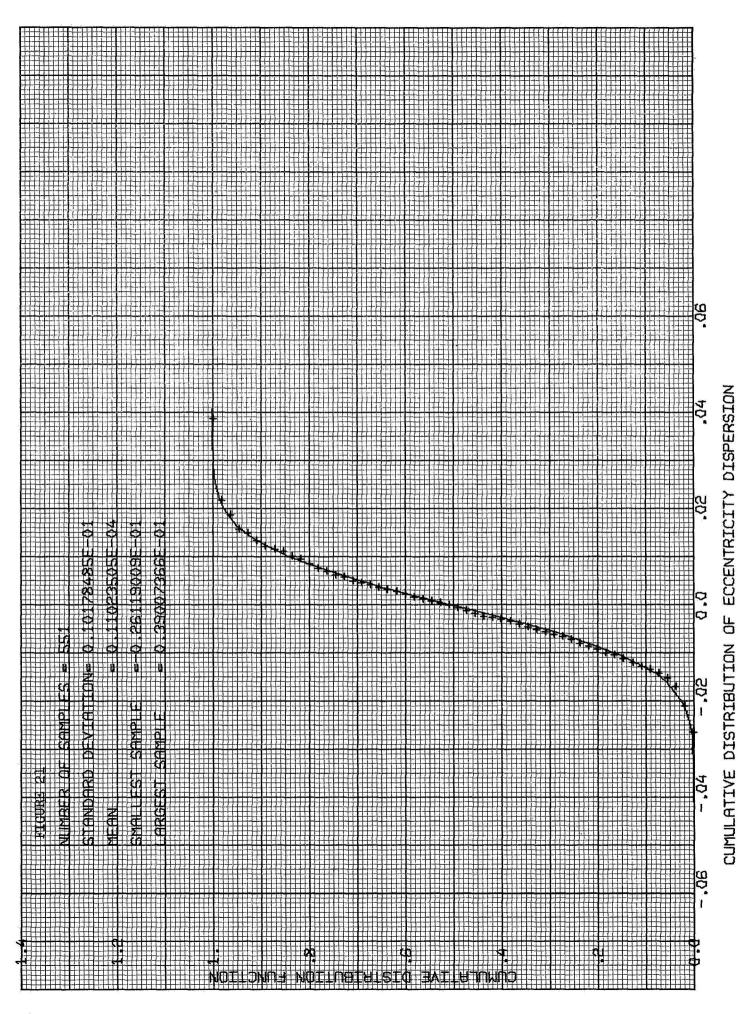
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-1.779798E-02 9.547430E-03 6.867104E-02 4.782530E-02	000	-3.264617E-04 1.193992E-01 4.183384E-03	-844578E- -453578E- -227011E-	1.428385E-02 1.242486E-02 -6.332770E-02	5.886191E-01) -2.027712E-04	-1.053953E-0D 1.719607E-03 1.636844E-02 1.443731E-03 -3.086392E-04 5.725628E-01)
-5-437500E 00 -1-487793E 00 2-716650E 01 2-444336E 01	.0	-1.606445E-01 4.194727E 01 -6.987305E-01		2.011230E 00 1.411621E 00 -1.740967E 01	6.987598E 01 -5.453613E 00 -5.179199E 00	-5.266406E 01 4.894092E 01 -2.740723E 00 -9.379883E-01 -1.977344E 01 -2.571143E 01
+2.474500E 03 1.354250E 03 8.682250E 03 7.656250E 03	The second secon	-4.075000E 01 1.486050E 04 6.072500E 02		1.690000E 02 1.510000E 02 -7.870000E 02	1.5	775000E 875000E 925000E 150000E 500000E
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86. TMY2 87. PM30 88. ISP3 89. NFR3 90. KRP3	1000	94. R0E3 95. TMP3 96. TMY3		101 TMY4 101 TMY4 102 M4CP		106 MFR5 107 ISP5 108 TMP5 109 TMY5 L10 MSCP

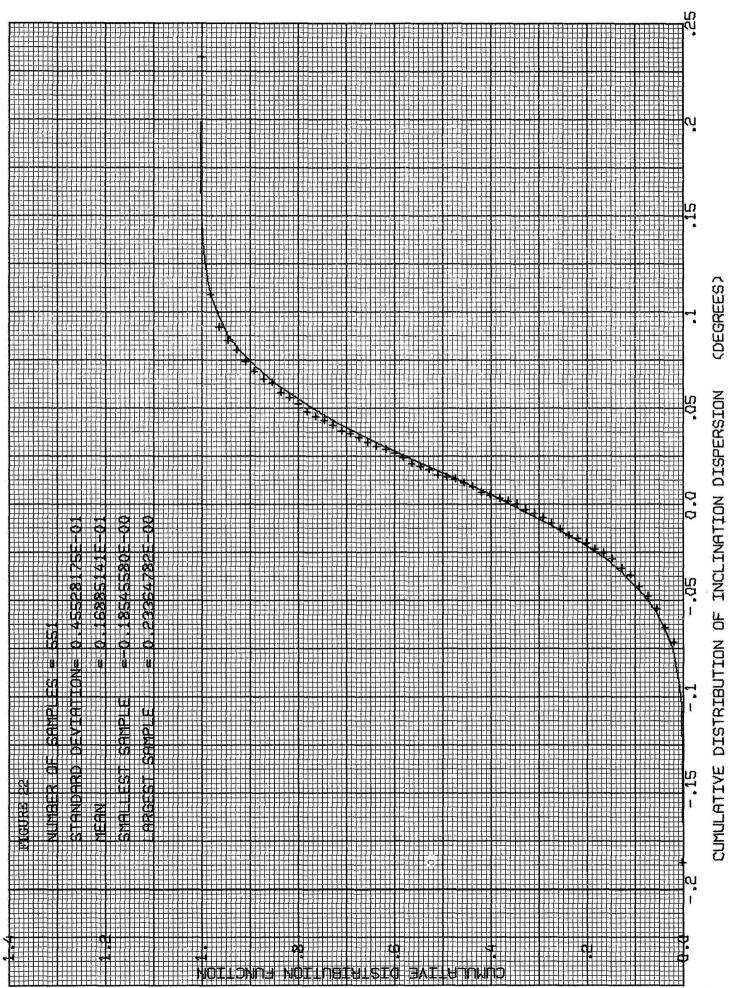
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MPI	1.6 70000 E-03	1.473200E	- 40			.874600		.698900	,4	8.299600E 0	.375100	00
	,	.821600E	20		04	3.916400E	02	5.132600E	100	-5.533100E 01	1.583400E	000
			100	.6.638000E		.092600		.229600		2.777039E 0	.541400	0
		700E	}	3		.185400		•459100		5.553900E 0	.708300	0.0
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8		1	3	4.877800E		.940300	03	.056500	~	1.062300E-01	2.767500	~
		283800E		3.153300E		•643200	03	.051700		.406500E-0	.816200	-4
		-4.567700E	600	6.306600E	700	1.328600E	400	-2.103300E	100	812900E	6323	D/
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		-5.103400E O	4	2.129700E		.695800	N	2.609500		.094400E 0	.214400	Ő
2		-3.519100E	.	1.485500E	40	3.942500E	02	-1.903500E	02	6.896030E 01	2.271100E	ROI
		-1.714700E (\	7.719800E		.089100		.499000		.427600E 0	.215700	0
		1.752600E	*	7.162400E		.069700		.420500		.208500E 0	.132000	0
		3.586300E	4	1.406700E		.387600		.938800		.398500E 0	.386000	0
3	The same of the sa	5.238400E	04			.122700		.702500		.006800E 0	.377600	0
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			2	2.338600E		.146300	ĺ	3.846500	0	1.338500E 0	.618200	-
		-1.594500E	03	-4.728700E	02	2.283100E	70	-7.786900E	00	-2.517700E 00	1.140300E	02
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			i	1.517500E		.207200		.890500		.120800E 0	.050700E	0
9			03	1.652900E		.212000		2.753100		.898500E 0	.937690E	0
				3.262900E		.991800		.504000		.919800E 0	.060600E	0
			04	4.976800E		.518400		8.253900		.853000E 0	.644400E	0
HY3	5.5700C0 E-04	•										
		57300E	- 20	2.	05	8.560100		001160	0	1.390900E 00	5.027700	
		30400E	•			.373200		.267800	0	1.234800E-01	.565400	
			1		7	3.32210	60	1.187700E	00	10-3000699	-1.921100E	10
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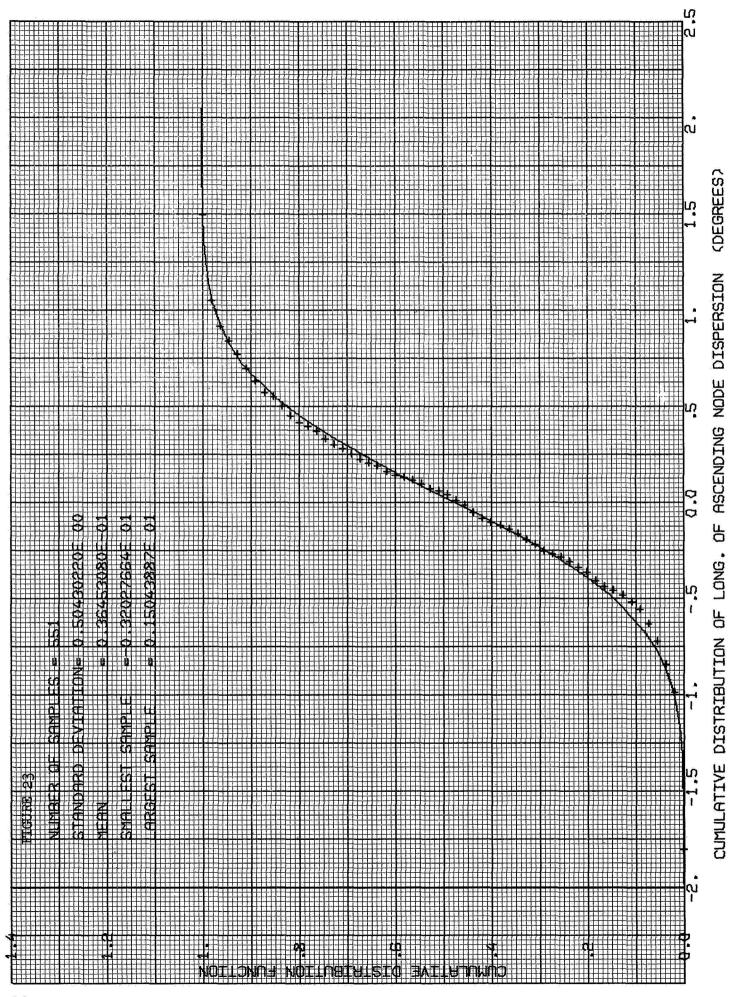
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		0	-4.7	56700E	-4	1.493000	3	.525400E	_	3.15380 CE		1.006500	
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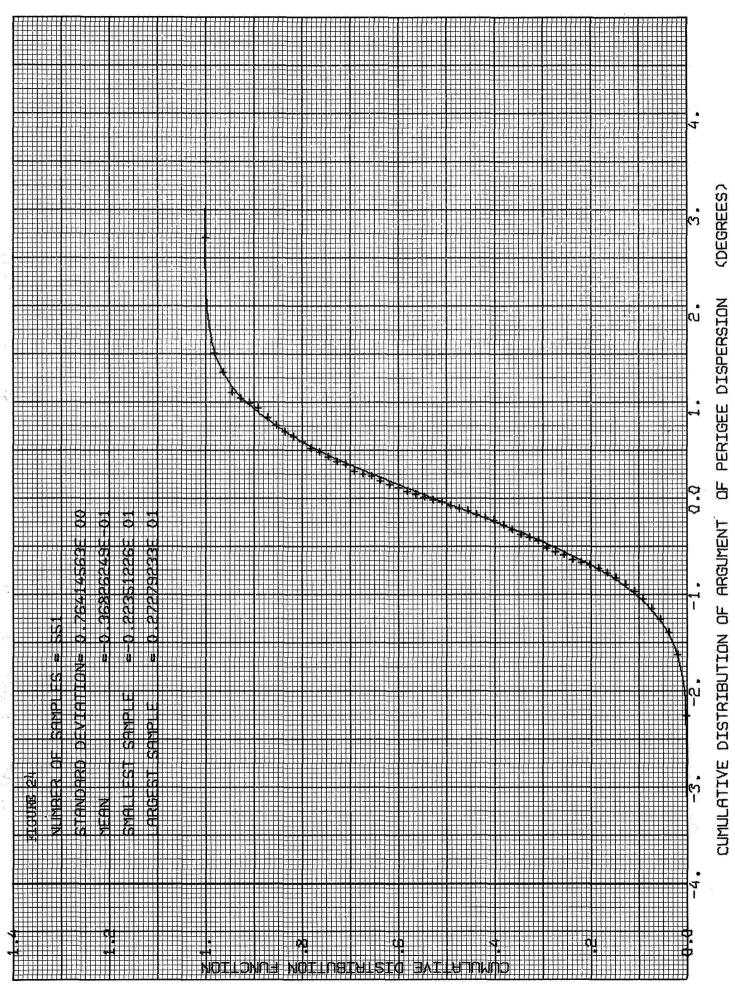
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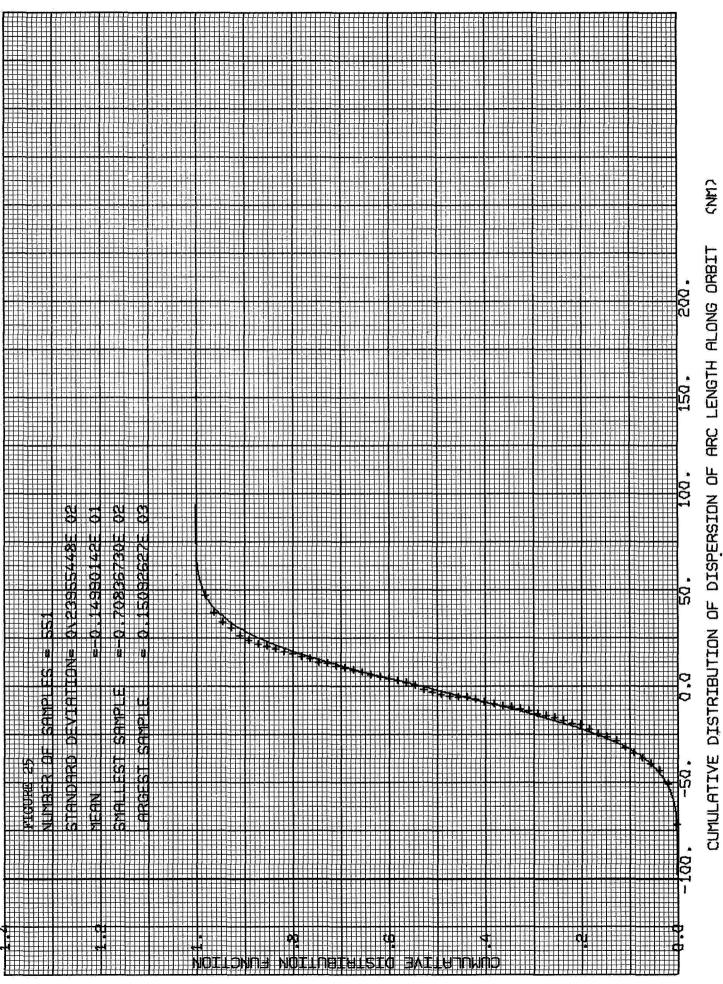


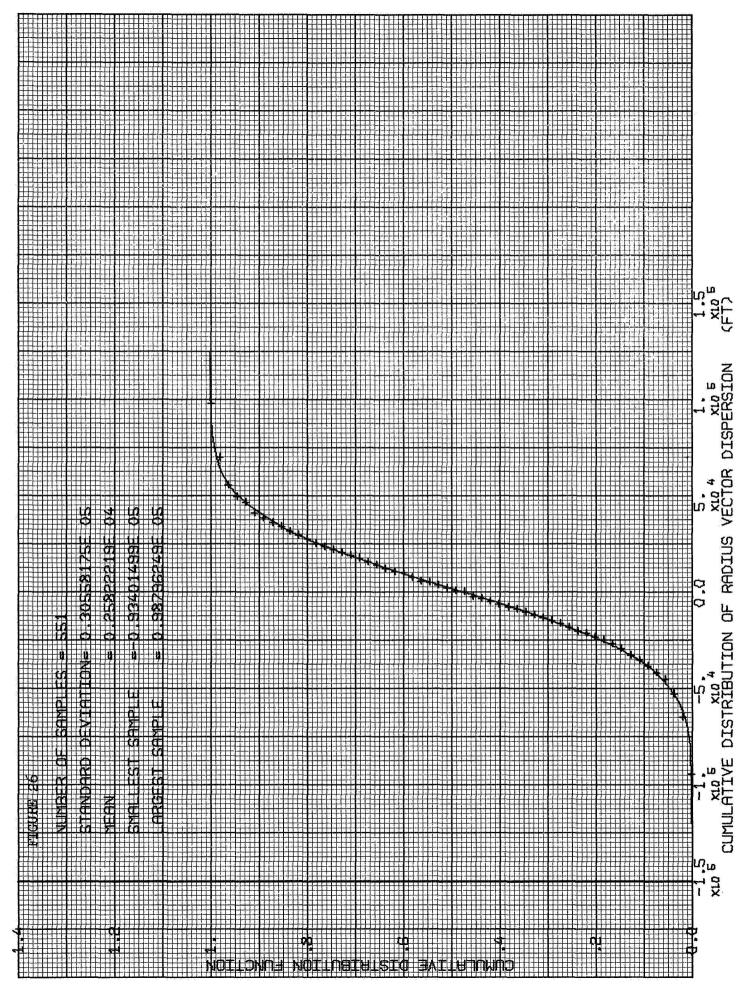


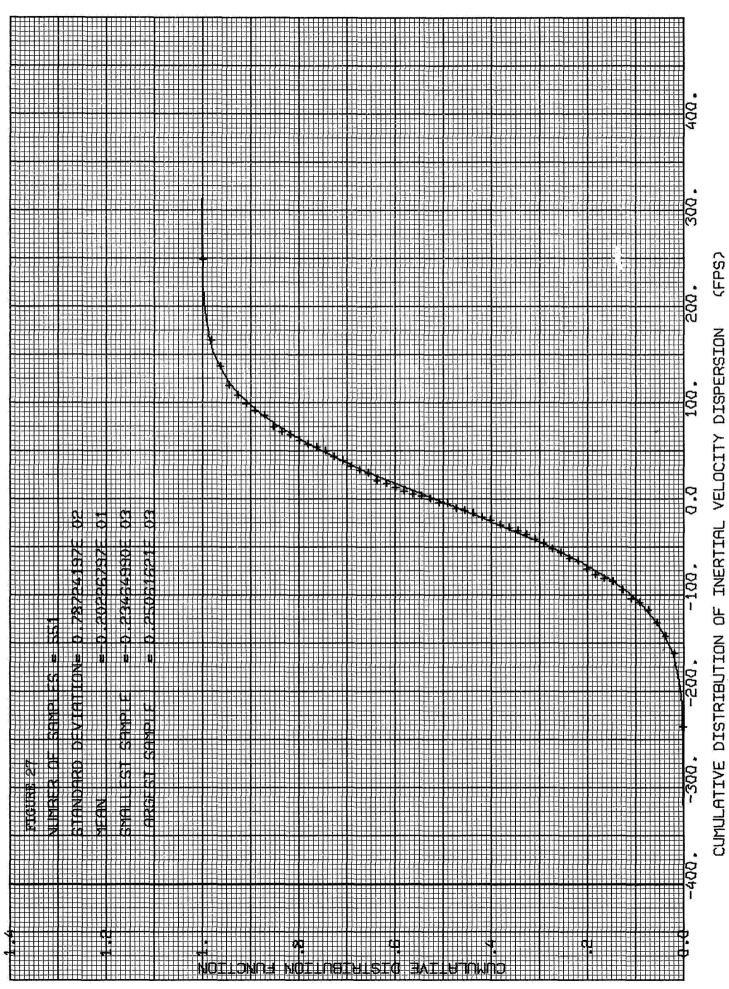


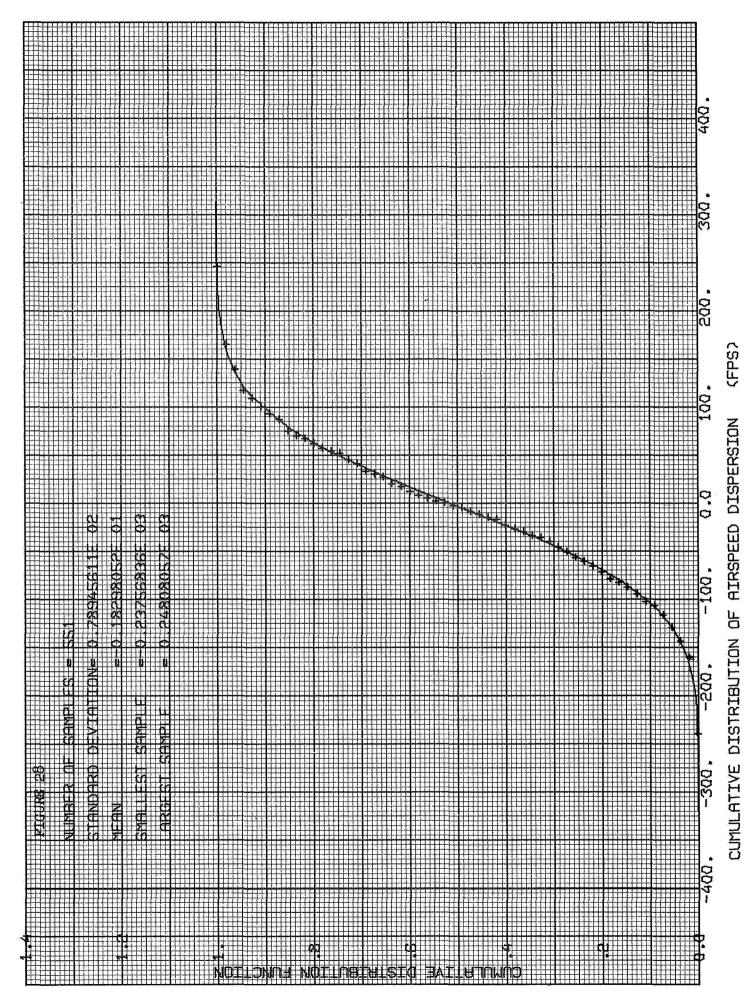


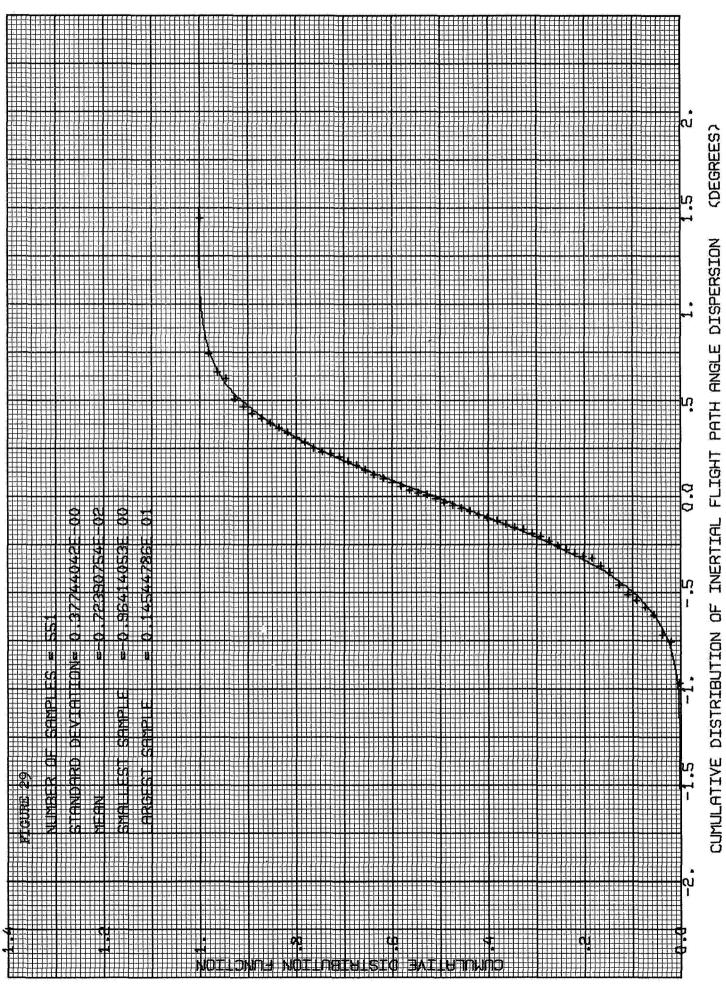


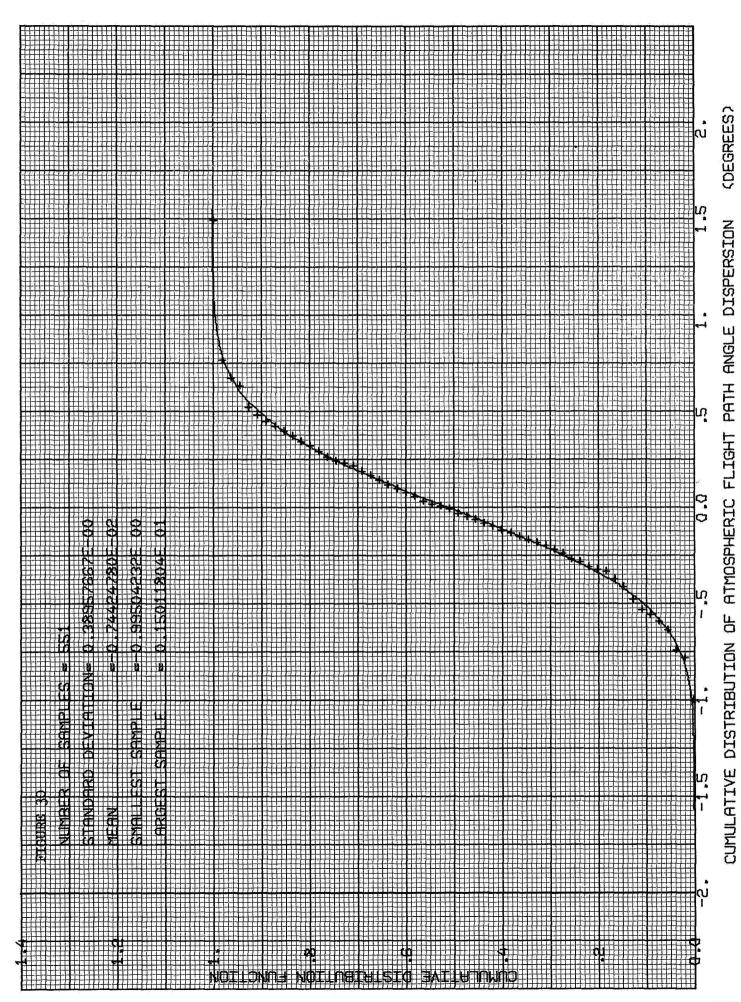


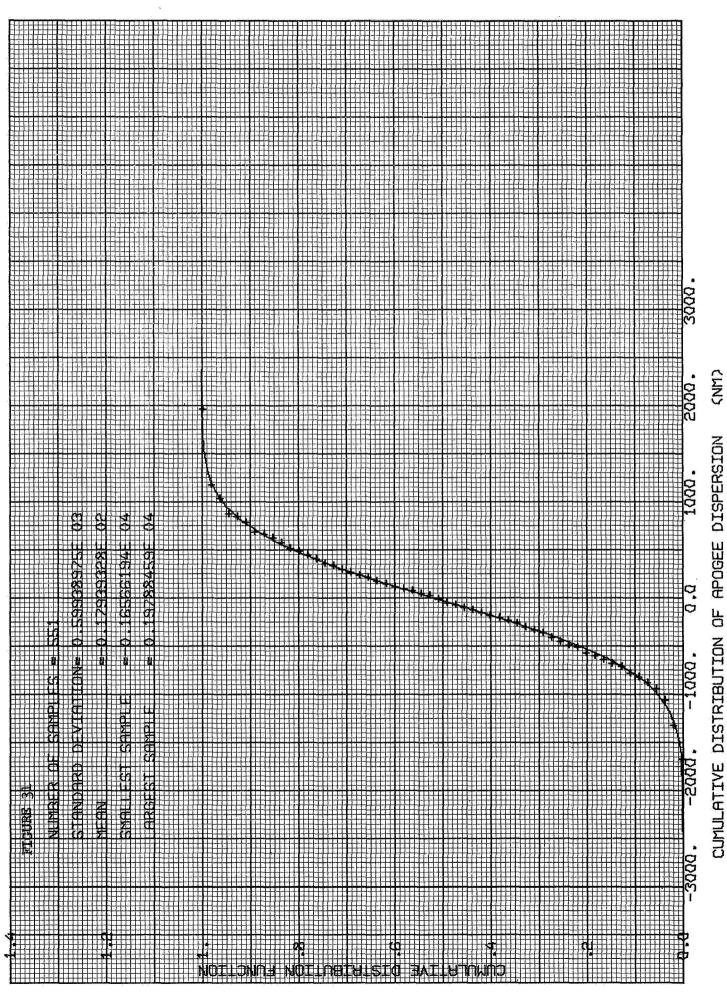


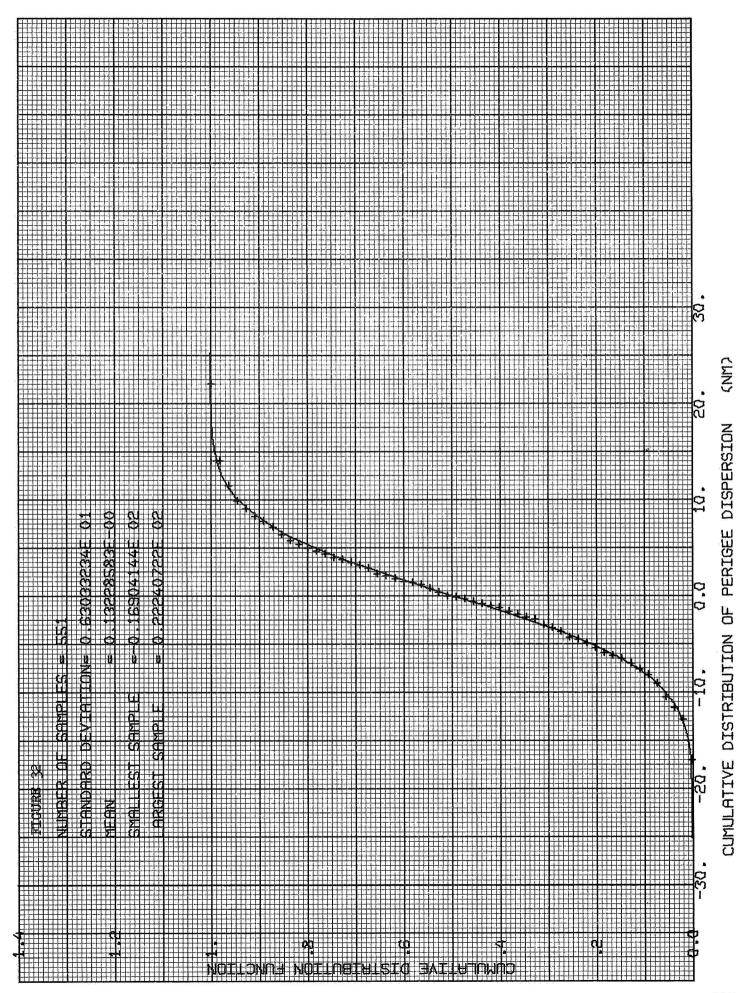


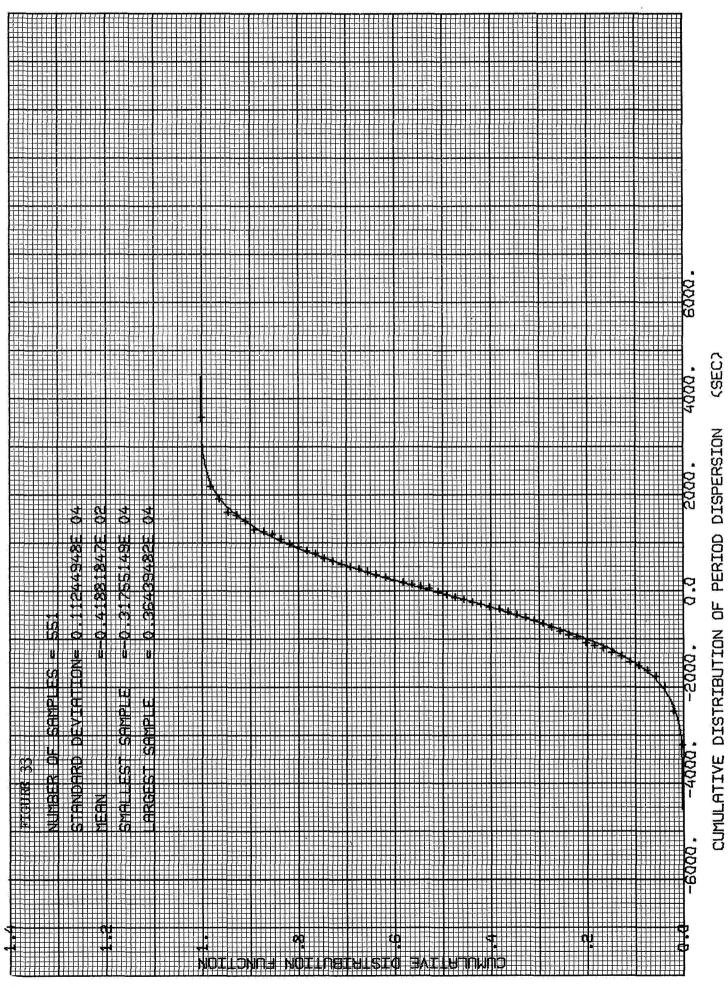


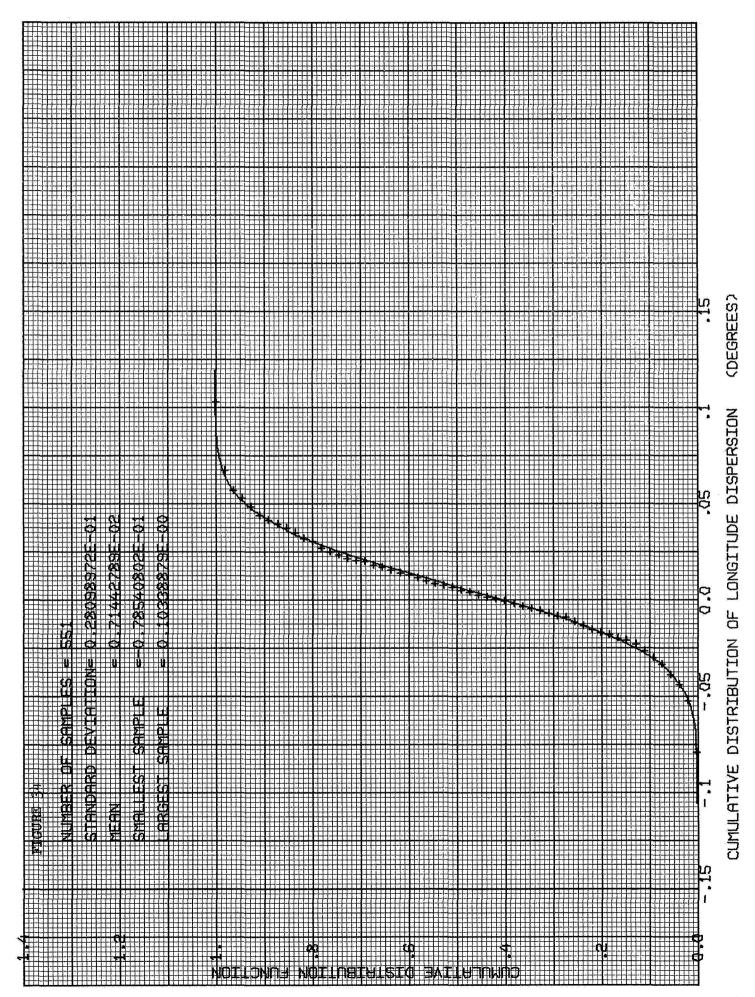


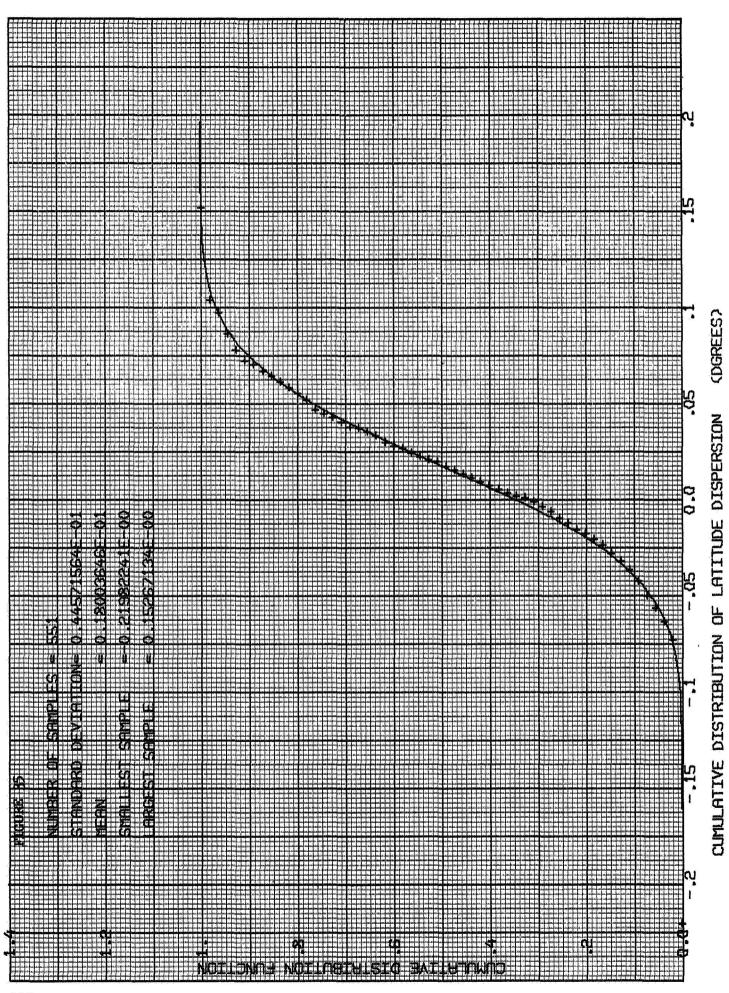


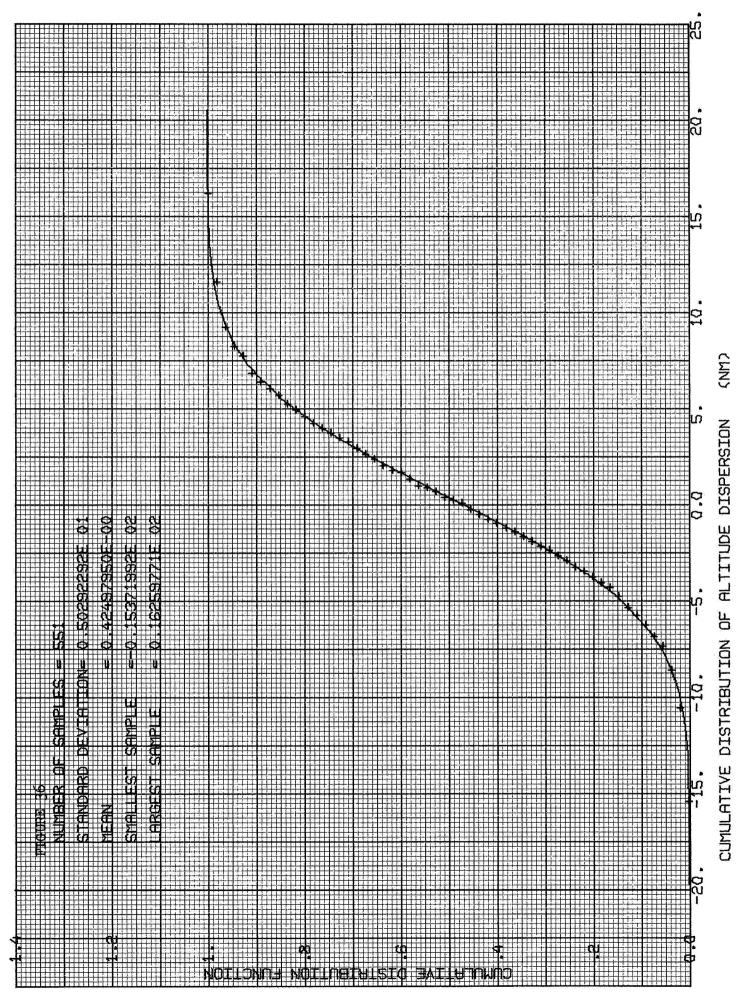












CONCLUSIONS

The results of the analysis of two missions are presented in this report.

These results indicate that, although the overall accuracy is not radically chang the effects of any individual error source is strongly dependent on the particula mission trajectory which is used.

Some general comments comparing the reentry mission, excape mission and the polar orbital mission which was analyzed in Phase I of this contract are in order The least sensitive trajectory is the excape mission since the large fifth stage burn reduces the effects of the lower stages, particularly in inclination, while burn itself is of such short duration that the trajectory errors are small.

The most sensitive is the reentry trajectory, partly due to the fact that the errors are propagated from burnout to reentry but also due to an increased fo stage sensitivity.

REFERENCES

1. "Scout Error Analysis Phase I Final Report" by L.B. Cohen, etal., NASA CR 66596.